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Aerodynamic Influence Coefficients
from Slender-Body Theory:
Analytical Development
and Computational Procedure

31 OCTOBER 1962

Prepared by WILLIAM P. RODDEN and EDITH F. FARKAS

Aeromechanics Department

Aerodynamics and Propulsion Research Laboratory

and

GEORGE Y. TAKATA

Computation and Data Processing Center

Laboratories Division

Prepared for COMMANDER SPACE SYSTEMS DIVISION
UNITED STATES AIR FORCE

Inglewood, California

<u>A</u>

LABORATORIES DIVISION • AFROSPACIE CORPORATION CONTRACT NO. AF 04(695)-169



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AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY: ANALYTICAL DEVELOPMENT AND COMPUTATIONAL PROCEDURE

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William P. Rödden and Edith F. Farkas
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ABSTRACT

A method is reviewed for computing the aerodynamic influence coefficients (AICs) for slender bodies. The method is based on the unsteady slender-body theory by Miles and its extension to obtain the AICs by Rodden and Revell.

The simplicity of the slender-body theory permits the definition of a number of sets of AICs for use in transient analysis. The influence coefficients relating the transient aerodynamic forces to the body deflections and their first two derivatives are defined by the following relation:

$${F(t)} = (qS/\overline{c})([C_{hs}] h + [C_{hd}] h \overline{c}/v + [C_{hi}] (\overline{h}\overline{c}^2/v^2)$$

The matrices $[C_{hs}]$, $[C_{hd}]$, and $[C_{hi}]$ are seen to be steady, damping, and inertial AICs, respectively. The oscillatory AICs are defined by

$$\{F\} = \rho \omega^2 b_r^2 s[C_h]\{h\}$$

and are related to the above definitions through

$$2k_{r}^{2}(\overline{c}s/S)[C_{h}] = [C_{hs}] + ik_{r}(\overline{c}/b_{r})[C_{hd}] - k_{r}^{2}(\overline{c}/b_{r})^{2}[C_{hi}]$$

The Aerospace IBM 7090 Computer Program No. HM15 provides the AICs in printed and optional punched-card output formats. The program capacity is 50 control points and, in the oscillatory case, 50 values of reduced velocity.

CONTENTS

ABST	RACT.		ii
SYMBO	OLS .		v
I.	FORM	MULATION OF PROBLEM	1
	A	Introduction	1
	В.	Sign Convention	1
	C.	Derivation of Equations	1
	D.	References	12
II.	GENI	ERAL DESCRIPTION OF INPUT	13
	Α.	Units	13
	в.	Classes of Numerical Data and Limitations	13
III.	DATA	A DECK SETUP	15
	A	Loading Order	15
	B.	Input Data Description	15
	C.	Example Keypunch Forms	17
IV.	PROC	GRAM OUTPUT	21
	A	Printed Output	21
	B.	Punched Output	27
v.	PROC	CESSING INFORMATION	29
	A	Operation	29
	В.	Estimated Machine Time	29
	C.	Machine Components Used	29
VI.	PRO	GRAM NOTES	31
	A	Subroutines Used	31
	В.	Generalized Tapes	31

CONTENTS (Continued)

VII.	FLOW DIAGRAM	33
VIII.		35
	FIGURES	
1.	Slender-body geometry for AICs	3
2.	Cross section of finned region of slender body	11
3.	Five-segment body	13

SYMBOLS

b _r	Reference semichord
C_{h}	Element of oscillatory AIC matrix
C _{hd}	Element of damping AIC matrix
C _{hi}	Element of inertial AIC matrix
C _{hs}	Element of steady AIC matrix
ਣ	Reference chord
F	Control point force
h	Control point deflection
I	Momentum of cross-flow virtual mass
k _r	Reference reduced frequency, $k_r = \omega b_r/V$
q	Free stream dynamic pressure
R	Body radius in finned region
S	Reference area
S(x)	Body cross-sectional area
S _{n-1/2} , S _{n+1/2}	Cross-sectional area of aft and forward ends, respectively, of n'th body segment
s	Reference semispan
t	Time
v	Free stream velocity
w	Downwash
x, y, z	Cartesian coordinates
$\Delta_{\mathbf{n}}$	Length of n'th body segment

SYMBOLS (Continued)

ΔV_n	Volume of n'th body segment
ρ	Free stream density
ω	Circular frequency
[]	Square matrix
	Column matrix

SECTION I FORMULATION OF PROBLEM

A. Introduction

The simplest theory available for the estimation of unsteady aerodynamic loads on slender bodies is the momentum theory of Munk as extended to the unsteady case by Miles. The limitations of the so-called slender-body theory have also been summarized by Miles (Ref. 4, Table 2, p. 161). These are: (1) the fineness ratio must be much less than unity; (2) the Mach number must be much less than the reciprocal of the fineness ratio; and (3) the motion must be slowly varying, or, more specifically, in the oscillatory case the reduced frequency must be of order unity. Hence, we see that the slender-body theory can be useful in the supersonic flight regime provided that the body is sufficiently slender and the motion is not violent.

The present formulation is based on the derivation of the oscillatory aerodynamic influence coefficients (AICs) from slender-body theory given by Rodden and Revell. The simplicity of slender-body theory permits the extension of Ref. 5 to obtain a number of sets of AICs for use in transient analysis, and this extension is made in the present treatment.

B. Sign Convention

The standard NASA stability axis system is used throughout. The positive directions of x, y, and z are forward, starboard, and downward, respectively. Positive rotations are given by applying the right-hand rule to the coordinate directions. The vehicle is assumed to be moving in the positive x-direction; i. e., the relative wind is in the negative x-direction. The force sign convention is the same as the coordinate convention.

C. Derivation of Equations

The derivation of equations for the oscillatory case is given in Ref. 5 (pp. 60-72). However, Ref. 5 contains a mistake in sign that leads to the incorrect sign on the damping terms, so it becomes necessary to repeat the

derivation here with the correction. We shall present the derivation for the oscillatory case since, by properly identifying the various terms, we may indicate the results for the transient case. We define the transient influence coefficients to relate the aerodynamic forces to the body deflections and their first two derivatives by the following

$$\{F(t)\} = (qS/\bar{c}) \left([C_{hs}] \{h\} + [C_{hd}] \{h\bar{c}/V\} + [C_{hi}] \{h\bar{c}^2/V^2\} \right) . \tag{1}$$

The matrices [Chs], [Chd], and [Chi] are seen to be steady, damping, and inertial AICs, respectively. The oscillatory AICs are defined by

$$\{F\} = \rho \omega^2 b_r^2 s [C_h] \{h\} \qquad (2)$$

Equations (1) and (2) must be identical in the oscillatory case, so the following relationship must exist among the various AICs

$$2k_{r}^{2}(\bar{c}_{s}/S)[C_{h}] = [C_{hs}] + ik_{r}(\bar{c}/b_{r})[C_{hd}] - k_{r}^{2}(\bar{c}/b_{r})^{2}[C_{hi}] .$$
 (3)

The equations given by Bisplinghoff, Ashley, and Halfman⁶ provide a convenient basis for deriving the AICs of a slender body. The vertical force acting per unit length of the body is the reaction to the substantial rate of change of the momentum of the virtual mass per unit length of the body

$$\frac{\mathrm{dF}}{\mathrm{dx}} = -\frac{\mathrm{D}}{\mathrm{Dt}} \left(\frac{\mathrm{dI}}{\mathrm{dx}} \right) \tag{4}$$

where the substantial derivative operator is given by

$$\frac{D}{Dt} = -V \frac{\partial}{\partial x} + \frac{\partial}{\partial t}$$
 (5)

since the relative wind is moving in the negative x-direction (see Fig. 1).

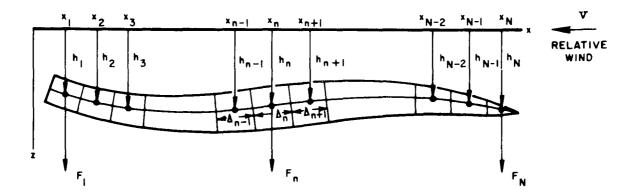


Fig. 1. Slender-body geometry for AICs.

The momentum of the virtual mass per unit length is found from the local cross-sectional area and the downwash

$$\frac{dI}{dx} = \rho S(x)w(x, t)$$
 (6)

where the downwash is the substantial derivation of the deflection

$$w(x, t) = \frac{Dh}{Dt}$$
 (7a)

$$= -V \frac{\partial h}{\partial t} + \frac{\partial h}{\partial t} \qquad . \tag{7b}$$

If we assume harmonic motion and substitute Eqs. (5), (6), and (7b) into Eq. (4), we obtain

$$\frac{dF}{dx} = \rho V \frac{d}{dx} \left\{ S(x) \left(-V \frac{dh}{dx} + i\omega h \right) \right\} - i\omega \rho S(x) \left(-V \frac{dh}{dx} + i\omega h \right) \qquad (8)$$

To obtain the force on a specified length of the body, it is necessary to integrate Eq. (8) over that length. We consider the body to be divided into a

number of sections, not necessarily of equal length, as shown in Fig. 1. For the n'th section, the control point is taken at the midpoint of its length Δ_n . The aft end of the section is located at $x_{n-1/2} = x_n - \Delta_n/2$ and has the crosssection area $S_{n-1/2}$; the forward end of the section is located at $x_{n+1/2} = x_n + \Delta_n/2$ and has the cross-section area $S_{n+1/2}$. Carrying out the integration of Eq. (8) for the n'th section yields the following

$$F_{n} = \int_{x_{n-1/2}}^{x_{n+1/2}} \frac{dF}{dx} dx \tag{9a}$$

$$= \rho V \left[S(x) \left(-V \frac{dh}{dx} + i\omega h \right) \right]_{x_{n-1/2}}^{x_{n+1/2}} - i\omega \rho \int_{x_{n-1/2}}^{x_{n+1/2}} S(x) \left(-V \frac{dh}{dx} + i\omega h \right) dx \qquad (9b)$$

$$= \rho V \left[-V(S_{n+1/2} h'_{n+1/2} - S_{n-1/2} h'_{n-1/2}) + i\omega(S_{n+1/2} h_{n+1/2} - S_{n-1/2} h_{n-1/2}) \right]$$

$$- i\omega\rho \int_{x_{n-1/2}}^{x_{n+1/2}} S(x)(-Vh' + i\omega h) dx . \qquad (9c)$$

7

We resort to Lagrangian interpolation for the evaluation of the terms in Eq. (9c). For the first two terms, we choose parabolic interpolation

$$h = h_{n-1} \frac{(x - x_n)(x - x_{n+1})}{(x_{n-1} - x_n)(x_{n-1} - x_{n+1})} + h_n \frac{(x - x_{n-1})(x - x_{n+1})}{(x_n - x_{n-1})(x_n - x_{n+1})} + h_{n+1} \frac{(x - x_{n-1})(x - x_n)}{(x_{n+1} - x_{n-1})(x_{n+1} - x_n)} .$$
 (10)

The input to the program must be the coordinates of the endpoints (aft and forward) of each section, from which the control point locations are calculated by averaging. With a given total body length, it is not possible to specify the control point locations a priori.

For evaluation of the integral, we use only linear interpolation

$$h = h_n + h'_n(x - x_n)$$
 (11)

and

$$S(x) = (1/\Delta_n) \left[S_{n-1/2}(x_{n+1/2} - x) + S_{n+1/2}(x - x_{n-1/2}) \right] \qquad (12)$$

The three-point interpolation leads to three elements in each row of the AIC matrix. Therefore, from the definition of Eq. (2),

$$F_{n} = \rho \omega^{2} b_{r}^{2} s(C_{hn, n-1} h_{n-1} + C_{hn, n} h_{n} + C_{hn, n+1} h_{n+1}) \qquad (13)$$

By identifying Eq. (13) and Eq. (9c), after evaluating the integral of Eq. (9c) by means of Eqs. (11) and (12), we obtain the following relation for the AICs

$$C_{\text{hn}, n-1} h_{n-1} + C_{\text{hn}, n} h_n + C_{\text{hn}, n+1} h_{n+1} = (1/k_r^2 s) \left\{ (S_{n-1/2} h'_{n-1/2} - S_{n+1/2} h'_{n+1/2}) \right\}$$

+
$$i(k_{r}/b_{r})(S_{n+1/2}h_{n+1/2} - S_{n-1/2}h_{n-1/2} + h'_{n}\Delta V_{n})$$

- $(k_{r}^{2}/b_{r}^{2})[(\Delta_{n}^{2}/12)(S_{n-1/2} - S_{n+1/2})h'_{n} - h_{n}\Delta V_{n}])$, (14)

where ΔV_n is the volume of the n'th section.

If we evaluate the deflections in Eq. (10) at stations $x_{n-1/2}$ and $x_{n+1/2}$, and then, by differentiating Eq. (10), evaluate the slopes at stations $x_{n-1/2}$, x_n , $x_{n+1/2}$, then we may place the right-hand side of Eq. (14) in terms of the control point deflections h_{n-1} , h_n , and h_{n+1} . If we carry out this substitution,

the AICs are found by identifying the coefficients of the control point deflections on both sides of Eq. (14). The following results are obtained

$$C_{\text{hn, n-1}} = \left(\frac{1}{k_{r}^{2}} s D_{n-1} \right) \left\{ 2 S_{n+1/2} (\Delta_{n+1} - \Delta_{n}) - 2 S_{n-1/2} (3 \Delta_{n} + \Delta_{n+1}) + i (k_{r}/b_{r}) \right\}$$

$$\times \left[- S_{n+1/2} \Delta_{n} \Delta_{n+1} - S_{n-1/2} \Delta_{n} (2 \Delta_{n} + \Delta_{n+1}) - 2 \Delta V_{n} (\Delta_{n} + \Delta_{n+1}) \right]$$

$$- \left(\frac{k_{r}^{2}}{b_{r}^{2}} \right) \left(\Delta_{n}^{2}/6 \right) (S_{n+1/2} - S_{n-1/2}) (\Delta_{n} + \Delta_{n+1}) \right\} , \qquad (15)$$

where

$$D_{n-1} = (\Delta_{n-1} + \Delta_n)(\Delta_{n-1} + 2\Delta_n + \Delta_{n+1}) \qquad ; \tag{16}$$

$$C_{hn, n} = \left(1/k_{r}^{2} s D_{n}\right) \left[2 S_{n+1/2} (\Delta_{n-1} + 2 \Delta_{n} - \Delta_{n+1}) - 2 S_{n-1/2} (\Delta_{n-1} - 2 \Delta_{n} - \Delta_{n+1}) + i (k_{r}/b_{r}) \left[S_{n+1/2} \Delta_{n+1} (\Delta_{n-1} + 2 \Delta_{n}) - S_{n-1/2} \Delta_{n-1} (2 \Delta_{n} + \Delta_{n+1}) + 2 \Delta V_{n} (\Delta_{n+1} - \Delta_{n-1})\right] - \left(k_{r}^{2}/b_{r}^{2}\right) \left[\left(\Delta_{n}^{2}/6\right) (S_{n-1/2} - S_{n+1/2}) (\Delta_{n+1} - \Delta_{n-1}) - D_{n} \Delta V_{n}\right],$$

$$(17)$$

where

$$D_{n} = (\Delta_{n-1} + \Delta_{n})(\Delta_{n} + \Delta_{n+1}) \qquad ; \tag{18}$$

$$C_{\text{hn, n+1}} = \left(1/k_{\text{r}}^{2} \text{sD}_{\text{n+1}}\right) \left(2S_{\text{n-1/2}}(\Delta_{\text{n-1}} - \Delta_{\text{n}}) - 2S_{\text{n+1/2}}(\Delta_{\text{n-1}} + 3\Delta_{\text{n}}) + i(k_{\text{r}}/b_{\text{r}})\right)$$

$$\times \left[S_{\text{n+1/2}} \Delta_{\text{n}}(\Delta_{\text{n-1}} + 2\Delta_{\text{n}}) + S_{\text{n-1/2}} \Delta_{\text{n-1}} \Delta_{\text{n}} + 2\Delta V_{\text{n}}(\Delta_{\text{n-1}} + \Delta_{\text{n}})\right]$$

$$- \left(k_{\text{r}}^{2}/b_{\text{r}}^{2}\right) \left(\Delta_{\text{n}}^{2}/6\right) \left(S_{\text{n-1/2}} - S_{\text{n+1/2}}\right) \left(\Delta_{\text{n-1}} + \Delta_{\text{n}}\right) , \qquad (19)$$

where

$$D_{n+1} = (\Delta_{n-1} + 2\Delta_n + \Delta_{n+1})(\Delta_n + \Delta_{n+1})$$
 (20)

The above expressions are applicable for all intermediate sections of the body, sections which are centrally located as far as the interpolation and differentiation are concerned. The exceptions are the first and N'th sections. The counterpart of Eq. (14) for the first section is

$$C_{h1,1}h_{1} + C_{h1,2}h_{2} + C_{h1,3}h_{3} = (1/k_{r}^{2}s) |S_{1/2}h_{1/2}' - S_{3/2}h_{3/2}' + i(k_{r}/b_{r})$$

$$\times (S_{3/2}h_{3/2} - S_{1/2}h_{1/2} + h_{1}'\Delta V_{1}) - (k_{r}^{2}/b_{r}^{2}) |(\Delta_{1}^{2}/12) (S_{1/2} - S_{3/2})h_{1}' - h_{1}\Delta V_{1}|$$

$$(21)$$

Carrying out the evaluation of the appropriate deflections and slopes from Eq. (10) in terms of the first three control point deflections leads to the following coefficients for the first row of the AIC matrix

$$C_{h1,1} = \left(1/k_{r}^{2} s D_{1}\right) \left(2 S_{3/2} (3 \Delta_{2} + \Delta_{3}) - 2 S_{1/2} (4 \Delta_{1} + 3 \Delta_{2} + \Delta_{3}) + i(k_{r}/b_{r})\right)$$

$$\times \left[S_{3/2} \Delta_{2} (2 \Delta_{2} + \Delta_{3}) - S_{1/2} (2 \Delta_{1} + \Delta_{2}) (2 \Delta_{1} + 2 \Delta_{2} + \Delta_{3}) - 2 \Delta V_{1} (2 \Delta_{1} + 3 \Delta_{2} + \Delta_{3})\right]$$

$$- \left(k_{r}^{2}/b_{r}^{2}\right) \left[\left(\Delta_{1}^{2}/6\right) (S_{3/2} - S_{1/2}) (2 \Delta_{1} + 3 \Delta_{2} + \Delta_{3}) - D_{1} \Delta V_{1}\right], \quad (22)$$

where

$$D_1 = D_{n-1}$$
 with $n = 2$;

$$C_{h1,2} = \left(1/k_{r}^{2} s D_{2}\right) \left[2 S_{3/2} (\Delta_{1} - 2 \Delta_{2} - \Delta_{3}) + 2 S_{1/2} (3 \Delta_{1} + 2 \Delta_{2} + \Delta_{3}) + i(k_{r}/b_{r}) \right]$$

$$\times \left[S_{3/2} \Delta_{1} (2 \Delta_{2} + \Delta_{3}) + S_{1/2} \Delta_{1} (2 \Delta_{1} + 2 \Delta_{2} + \Delta_{3}) + 2 \Delta V_{1} (\Delta_{1} + 2 \Delta_{2} + \Delta_{3})\right]$$

$$- \left(k_{r}^{2}/b_{r}^{2}\right) \left(\Delta_{1}^{2}/6\right) (S_{1/2} - S_{3/2}) (\Delta_{1} + 2 \Delta_{2} + \Delta_{3}), \qquad (23)$$

where

$$D_2 = D_n \text{ with } n = 2$$
 ;

$$C_{h1,3} = (1/k_r^2 sD_3) \left[2S_{3/2}(\Delta_2 - \Delta_1) - 2S_{1/2}(3\Delta_1 + \Delta_2) + i(k_r/b_r) \left[-S_{3/2}\Delta_1\Delta_2 - S_{1/2}\Delta_1(2\Delta_1 + \Delta_2) - 2\Delta V_1(\Delta_1 + \Delta_2) \right] - (k_r^2/b_r^2) (\Delta_1^2/6) (S_{3/2} - S_{1/2})(\Delta_1 + \Delta_2) \right],$$
(24)

where

$$D_3 = D_{n+1}$$
 with $n = 2$.

Similarly, the counterpart of Eq. (14) for the N'th section is

$$C_{hN, N-2} h_{N-2} + C_{hN, N-1} h_{N-1} + C_{hN, N} h_{N} = \left(1/k_{r}^{2} s\right) \left[S_{N-1/2} h_{N-1/2}^{\prime} - S_{N+1/2} h_{N+1/2}^{\prime} + i(k_{r}/b_{r})(S_{N+1/2} h_{N+1/2} - S_{N-1/2} h_{N-1/2} + h_{N}^{\prime} \Delta V_{N}) - \left(k_{r}^{2}/b_{r}^{2}\right) \left[\left(\Delta_{N}^{2}/12\right) (S_{N-1/2} - S_{N+1/2}) h_{N}^{\prime} - h_{N} \Delta V_{N} \right] \right] , \qquad (25)$$

and we obtain the following coefficients for the last (N'th) row of the AIC matrix

$$C_{hN, N-2} = \left(1/k_r^2 s D_{N-2}\right) \left[2S_{N-1/2}(\Delta_{N-1} - \Delta_N) - 2S_{N+1/2}(3\Delta_N + \Delta_{N-1}) + i(k_r/b_r) \left[S_{N+1/2} \Delta_N(2\Delta_N + \Delta_{N-1}) + S_{N-1/2} \Delta_N \Delta_{N-1} + 2\Delta V_N(\Delta_N + \Delta_{N-1})\right] - \left(k_r^2/b_r^2\right) \left(\Delta_N^2/6\right) (S_{N-1/2} - S_{N+1/2})(\Delta_N + \Delta_{N-1})\right] , \qquad (26)$$

where

$$D_{N-2} = D_{n-1} \text{ with } n = N - 1$$

$$C_{hN, N-1} = \left(1/k_{r}^{2} s D_{N-1}\right) \left(2 S_{N+1/2} (3 \Delta_{N} + 2 \Delta_{N-1} + \Delta_{N-2}) + 2 S_{N-1/2} (\Delta_{N} - 2 \Delta_{N-1} - \Delta_{N-2})\right)$$

$$+ i(k_{r}/b_{r}) \left[-S_{N+1/2} \Delta_{N} (2 \Delta_{N} + 2 \Delta_{N-1} + \Delta_{N-2}) - S_{n-1/2} \Delta_{N} (2 \Delta_{N-1} + \Delta_{N-2})\right]$$

$$- 2 \Delta V_{N} (\Delta_{N} + 2 \Delta_{N-1} + \Delta_{N-2}) - (k_{r}^{2}/b_{r}^{2}) (\Delta_{N}^{2}/6) (S_{N+1/2} - S_{N-1/2}) (\Delta_{N} + 2 \Delta_{N-1} + \Delta_{N-2})$$

$$(27)$$

where

$$D_{N-1} = D_n \text{ with } n = N-1$$
;

$$C_{hN,N} = \left(1/k_{r}^{2} s D_{N}\right) \left[2 S_{N-1/2} (3 \Delta_{N-1} + \Delta_{N-2}) - 2 S_{N+1/2} (4 \Delta_{N} + 3 \Delta_{N-1} + \Delta_{N-2}) + i(k_{r}/b_{r}) \left[S_{N+1/2} (2 \Delta_{N} + \Delta_{N-1}) (2 \Delta_{N} + 2 \Delta_{N-1} + \Delta_{N-2}) - S_{N-1/2} \Delta_{N-1} (2 \Delta_{N-1} + \Delta_{N-2}) + 2 \Delta V_{N} (2 \Delta_{N} + 3 \Delta_{N-1} + \Delta_{N-2}) \right] - \left(k_{r}^{2}/b_{r}^{2}\right) \left[\left(\Delta_{N}^{2}/6\right) (S_{N-1/2} - S_{N+1/2}) + 2 \Delta V_{N} (2 \Delta_{N} + 3 \Delta_{N-1} + \Delta_{N-2}) - D_{N} \Delta V_{N}\right]\right],$$

$$\times (2 \Delta_{N} + 3 \Delta_{N-1} + \Delta_{N-2}) - D_{N} \Delta V_{N}\right],$$

$$(28)$$

where

$$D_{N} = D_{n+1} \text{ with } n = N - 1 \qquad .$$

To illustrate the assembly of the coefficients into the AIC matrix, we show the format below for a slender body having five degrees of freedom.

$$\begin{bmatrix} C_{h1,1} & C_{h1,2} & C_{h1,3} & 0 & 0 \\ C_{h2,1} & C_{h2,2} & C_{h2,3} & 0 & 0 \\ 0 & C_{h3,2} & C_{h3,3} & C_{h3,4} & 0 \\ 0 & 0 & C_{h4,3} & C_{h4,4} & C_{h4,5} \\ 0 & 0 & C_{h5,3} & C_{h5,4} & C_{h5,5} \end{bmatrix}$$
(29)

If we formally write each element of the oscillatory AIC matrix in the form

$$C_{h_{ij}} = (1/k_r^2 s D_j) [A_{ij} + i(k_r/b_r) B_{ij} - (k_r^2/b_r^2) C_{ij}]$$
, (30)

then the elements of the transient AIC matrices may be identified by comparison with Eq. (3). We see that the elements of the steady AIC matrix are given by

$$C_{hs_{ij}} = 2(\overline{c}/S) A_{ij}/D_{j} , \qquad (31)$$

the elements of the damping AIC matrix are given by

$$C_{hd_{ij}} = (2/S) B_{ij}/D_{j} , \qquad (32)$$

and, finally, the elements of the inertial AIC matrix are given by

$$C_{hi_{ij}} = (2/\bar{c}S) C_{ij}/D_{j}$$
 (33)

Before concluding this discussion, we note that, according to Miles (Ref 4, p. 169) the slender-body theory presented above may be applied to finned vehicles if an effective cross-section area in the finned region, whose geometry is shown in Fig. 2, is taken as

$$S_e = \pi(s^2 - R^2 + R^4/s^2)$$
 (34)

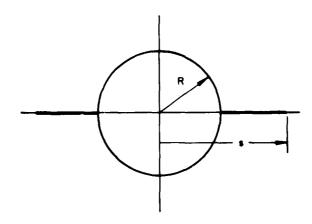


Fig. 2. Cross section of finned region of slender body.

D. References

- 1. M. M. Munk. "The Aerodynamic Forces on Airship Hulls." NACA Report No. 184, 1923.
- 2. J. W. Miles. "On Non-Steady Motion of Slender Bodies." Aeronautical Quarterly, 2 (1950), 183-194.
- 3. J. W. Miles. "Virtual Momentum and Slender Body Theory." Quarterly Journal of Mechanics and Applied Mathematics, 6 (1953), 286-289.
- 4. J. W. Miles. <u>The Potential Theory of Unsteady Supersonic Flow.</u> London: Cambridge University Press, 1959.
- 5. W. P. Rodden and J. D. Revell. "The Status of Unsteady Aerodynamic Influence Coefficients." Institute of the Aerospace Sciences S.M.F. Fund Paper No. FF-33, 23 January 1962.
- 6. R. L. Bisplinghoff, H. Ashley, and R. L. Halfman. <u>Aeroelasticity</u>. Reading: Addison-Wesley Publishing Co., Inc., 1955, p. 418.

SECTION II GENERAL DESCRIPTION OF INPUT

A. Units

Since all dimensional input is geometrical and the aerodynamic matrix is dimensionless, only a consistent set of length units is necessary: inches or feet.

B. Classes of Numerical Data and Limitations

The data required by the program are control indicators, geometry, and a set of reduced velocities. The example problem illustrates their use.

1. Example Problem

We consider the five-segment body shown in Fig. 3 computing the oscillatory case for the reduced velocities $(1/k_r)$ of 2.0 and 6.0, the transient case (steady, damping, and inertial AICs), and the steady case (steady AICs only).

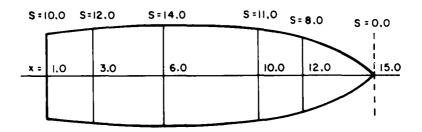


Fig. 3. Five-segment body.

The geometrical data required as input to the program are listed below.

Segment No.	\times (ft)	S (sq ft)
1 (aft)	1	10
1 (forward), 2 (aft)	3	12
2 (forward), 3 (aft)	6	14
3 (forward), 4 (aft)	10	11
4 (forward), 5 (aft)	12	8
5 (forward)	15	0

$$\tilde{c} = 10.0 \text{ ft}$$

$$b_r = 5.0 \text{ ft}$$

$$s = 20.0 \text{ ft}$$

2. Program Restrictions and Options

a. The maximum number of segments into which a body may be subdivided must be ≤ 50 .

S = 200.0 sq ft

- b. The maximum number of $1/k_{r}$'s for one data deck must be \leq 50.
- c. If it is desired to compute the steady case, a zero value of $1/k_{\rm r}$ must be used, to compute the transient case a negative value of $1/k_{\rm r}$ must be used.

SECTION III DATA DECK SETUP

A. Loading Order

Input data decks punched from keypunch forms are loaded behind column binary deck HM15. Any number of complete decks may be stacked. The data for each deck must be in the following order:

- (1) Heading card
- (2) IX, KVBRW, IPUNCH
- (3) CBAR, BR, S, SREF
- (4) X(I) series
- (5) LOCAL AREA(I) series
- (6) $(1/k_r)_i$ series

B. Input Data Description

- The heading card may contain any information desired in Columns
 through 72. Column 1 is always left blank.
- (2) Control card (FORMAT 1814)
 - (a) IX = the number of control points; also the number of body segments, ≤ 50
 - (b) KVBRW = the number of reduced velocities to be listed in data item (6), ≤ 50
 - (c) IPUNCH = 0 or blank if the computed matrices are to be punched in cards;
 - ≠ 0 if no punched output is desired.

(3) Single parameters card (FORMAT 6E12.8)

- (a) $CBAR = \bar{c}$, reference chord
- (b) BR = b, reference semichord
- (c) S = s, reference semispan
- (d) SREF = S, reference area

The constants are tabulated in the order in which they are defined. CBAR and SREF are used in the steady and transient cases; BR and S (semispan) are necessary for computing oscillatory cases. When only one pair of the constants is needed, the two respective card fields for the other pair may contain zero or be left blank.

(4) X(I) Segment coordinate series, $I \le 51$ (FORMAT 6E12.8)

The x coordinates $(x_{n-1/2} \text{ and } x_{n+1/2})$ used in this series locate the aft and forward end of each body segment. The number of terms in the series is one more than the number of control points (I = 1, IX + 1). Begin the series with $x_{n-1/2}$ (aft end) of the aft body segment and list the consecutive coordinates through $x_{n+1/2}$ (forward end) of the foremost body segment (nose).

(5) LOCAL AREA (I) series (FORMAT 6E12.8)

These are the local cross-section areas $(S_{n-1/2} \text{ and } S_{n+1/2})$ at the aft and forward end of each body segment. The number of terms in this series is the same as in X(I). The areas are listed in the same order as the coordinates; begin with $S_{n-1/2}$ for the aft body segment and list the consecutive areas through $S_{n+1/2}$ for the foremost body segment.

(6) $(1/k_r)_i$ series (FORMAT 6E12.8)

This series consists of the reference reduced velocities for the oscillatory case and the codes for obtaining the steady and transient cases. To obtain the steady matrix $[C_{hs}]$ input $1/k_r = 0.0$ and for the

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transient case ($[C_{hs}]$, $[C_{hd}]$, and $[C_{hi}]$) use $1/k_r$ = any negative number. The number of $1/k_r$'s in the series is set by KVBRW in the control card.

NOTE: Each new series starts on a new line (card).

C. Example Keypunch Forms

Example keypunch forms are given on the following pages. Columns 73 through 80 are reserved for data deck identification. This space may be used in any fashion; however, it is suggested that the last three columns be used for sequencing. In the example that follows, only the sequenced cards (lines) are to be used in the sample data deck; the lines with blank Columns 73 through 80 are for explanation of the input.

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SECTION IV PROGRAM OUTPUT

A. Printed Output

- 1. All input data.
- 2. The segment lengths [DELTA X (I)] and the control point coordinates XN (I), (Δ_n and x_n in Section I.C).
- 3. Each AIC matrix preceded by the associated $1/k_{_{\mbox{\scriptsize r}}}$ and the matrix size.
- 4. Sequencing number of the first and last punched cards (output) for each AIC matrix.
 - 5. Example problem printed output is given on the following pages.

HEADING CARD FOR USER OF AIC FROM SLENDER-BODY THEORY

AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY

INPUT DATA

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X(I)

X(1)		LOCAL AREA	
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0.59999999E 0	10	.13999999E	02
0.0999999E	02	.11000000E	02
OE)2	8000000E	01
0.15000000E 0)2	•0	

OUTPUT DATA

(I)NX	01001	000
CONTROL POINT XN(I)	0.20000000E 01 0.45000000E 01	0.11000000E 01 0.11000000E 02 0.13500000E 02
DELTAX(I)	0.2000000E 01 0.30000000E 01 0.40000000E 01	0.2000000E 01 0.3000000E 01
CONTROL PT. NO.	3 2 1	4 N

AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY

1./KR= 0.2000000E 01

NUMBER OF CONTROL POINTS= 5

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REAL	COLUMN 5	-0.37158729E-00 -0.79011904E 00 0.19765714E 01 -0.55121211E 00 -0.50303029E-01 COLUMN
IMAGINARY	COLUMN 4	0.59885713E 00 0.20800000E-00 -0.27428570E-00 0. COLUMN 10 0. 0. 0.16145454E-00 0.93090907E-01
REAL	COLUMN 3	0.73234285E 00 0.18154857E 01 -0.95718679E 00 0. 0. COLUMN 9 0. 0. 0.
IMAGINARY	COLUMN 2	-0.45599999E-00 -0.35799999E-00 0. 0. COLUMN 8 0. 0.3466666E-00 -0.1026666E-00
REAL	COLUMN 1	-0.31675555E-00 -0.94736665E 00 0. 0. COLUMN 7 0. -0.91938459E 00 0.10114666E 01 -0.52453332E 00

PUNCHED CARDS NOS. HM15 1 THRU HM15 13

AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY

1./KR=-0.30000000E 01

NUMBER OF CONTROL POINTS = 5

THE TRANSIENT CASE

COLUMN	COLUMN
COLUMN 5	COLUMN 5
0.	0.
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-0.21090908E-00	0.80727270E-01
0.29090908E-00	0.46545453E-01
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COLUMN 2	COEUMN 2
0.36571427E-00	0.29642856E-00
0.86857141E 00	0.10399999E-00
-0.47912087E-00	-0.13714285E-00
0.	0.
COLUMN 1	COLUMN 1
-0.179999996-00	-0.22799999E-00
-0.47333326-00	-0.17899999E-00
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PUNCHED CARDS NOS. HMI5 14 THRU HMI5 33

AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY

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NUMBER OF CONTROL POINTS = 5

THE STEADY CASE

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COLUMN 4	0°	0°	-0.45897435E-00		-0.2666666E-00
COLUMN 3	-0.18571428E-00	-0.39523809E-00	0.93809521E 00	-0.27575757E-00	-0.24242423E-01
COLUMN 2	0.36571427E-00	0.86857141E 00	-0.47912087E-00	0°	•0
COLUMN	-0.17999999E-00	-0.47333332E-00	•0	• 0	• 0

- 2645

PUNCHED CARDS NOS. HM15 34 THRU HM15 41

AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY

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NUMBER OF CONTROL POINTS= 5

THE OSCILLATORY CASE

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IMAGINARY	COLUMN	10857 16999 19714 15836 15272
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	72	01
REAL	Z W	-0.33430158E 01 -0.71139284E 01 0.16986094E 02 -0.49633332E 01 -0.43818181E-00
R	COLUMN	.334301 .711392 .169860 .496333 .438181
		. , , ,
<u>۲</u>	4	0.17965714E 01 0.62399999E 00 -0.82285713E 00 0. CCLUMN 10 0. 0.
IMAGINARY	COLUMN	1796571. 62399999 82285713 COLUMN 48436362
IMA	COL	0.179 0.623 0.822 0. 0. CGL 0. 0.
		1
	e.	13E 0 28E 0 36E 0 9E 0
REAL	COLUMN	0.65837713E 01 0.15712628E 02 0.86231206E 01 0. 0. COLUMN 9 0. 0. 0. 0. 0. 0. 0. 0. 0. 0.
	00	0.65837713E 01 0.15712628E 02 0.86231206E 01 0. COLUMN 9 0. 0. 0. 0. 0. 0. 0. 0. 0. 0.
	2	
NARY	z	-0.13679999E 01 0. 0. 0. COLUMN 8 0. 0. 0.10399999E 01 0.10399999E 01 0.30799999E 00
IMAGINARY	COLUMN	136799 107399 103999 307999 911999
н	0	
	-	7 7 7 01 01 01 01 01
ΑĹ	Z ¥	-0.31967555E 01 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.00 0
REAL	COLUMN	319675 852069 COLUMN 826297 879813 479119
		00000 0000
		110m4v 110m4v

PUNCHED CARDS NOS. HM15 42 THRU HM15 54

B. Punched Output

- 1. A deck of punched cards (output) from this program is suitable as an input deck to other programs requiring AICs.
- 2. All punched output is sequenced in order on Columns 73 through 80 starting with HM150001. The data is punched in the following order:
 - a. Card 1 contains (1/k_r): (FORMAT 6E12.8).
- b. Card 2 contains IX, the size (number of control points) of the AIC matrix (FORMAT 1814).
- c. The AIC matrix punched in column binary form and its TRA card make up the remainder of the punched output for $(1/k_r)$.
- 3. The order of Statement 2 can be repeated for all reduced velocities per input deck if we note that in the transient case, the three matrices $[C_{hs}]$, $[C_{hd}]$, and $[C_{hi}]$, which must be considered in Statement 2.c, are punched in the order listed with a TRA card after each matrix.
- 4. Each AIC matrix is punched by columns. Column 1 starts in Orgin 1 and Column 2 in Location (1 + matrix size).
- 5. The oscillatory matrices are punched in the order Column 1 (real), Column 1 (imaginary); Column 2 (real), Column 2 (imaginary); ...; Column IX (real), Column IX (imaginary). In the steady, damping, and inertial matrices all columns are real and are punched in order.

SECTION V PROCESSING INFORMATION

A. Operation

STANDARD FORTRAN MONITOR system

B. Estimated Machine Time

T = time in minutes

IX = number of control points

KVBRW = number of reduced velocities per deck

m = number of decks of input data

T = 0.5 + 0.01 $\sum_{j=1}^{m} (IX)_{j} (KVBRW)_{j}$

C. Machine Components Used

A number of core storages

Standard FORTRAN input tape (N1)

Standard FORTRAN output print tape (N2)

Standard FORTRAN output punch tape (N3)

SECTION VI PROGRAM NOTES

A. Subroutines Used

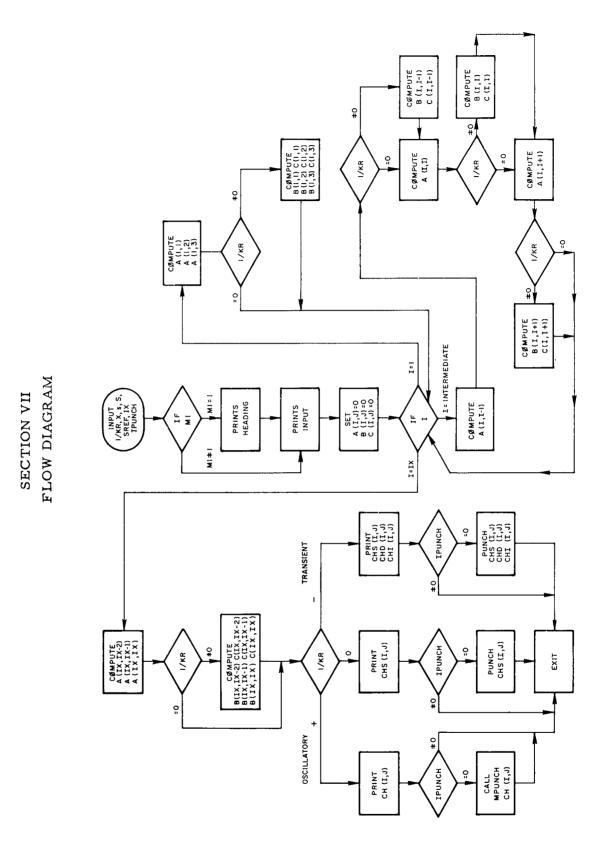
BINPU: binary punch routine

MPUNCH: punch AIC matrix

MPRINT: prints AIC matrix

B. Generalized Tapes

Input, print, and punch tapes are defined as Units 2, 3, and 5, respectively; however, these may be altered by placing the desired units on symbolic cards HM150007, HM150008, and HM150009.



-33-

SECTION VIII SYMBOLIC LISTING

Some of the symbols used in the program are defined as follows:

FORTRAN symbols	<u>Definition</u>
A (I, J).	$[C_{ ext{hs}}]$, also the associated working array
B (I, J)	$[C_{ m hd}]$, also the associated working array
BR	${f b}_{f r}$
CBAR	c
C (I, J)	$[C_{ m hi}]$, also the associated working array
CH (I, J), CH (I, J + 1)	Respective real and imaginary parts of the oscillatory AIC Matrix
DFI, DF2,, DF9	D_1 , D_2 , D_3 , D_{N-2} , D_{N-1} , D_N , D_{n-1} , D_n , and D_{n+1} , respectively
DELTAX (I)	Δ_{n} for segment i, i = n
DELVOL (I)	ΔV_n for segment i, i = n
IX	Number of control points
S	s
LOCAL AREA (I)	Used in printed output only, see SVAR (below)
SREF	S
SVAR (I)	$S_{n-1/2}$ for segment i (i = 1, IX) and $S_{n+1/2}$ for segment IX
SVMINU (I)	S _{n-1/2} for segment i
KVBRW	Number of reduced velocities included in
	the data deck

FORTRAN symbols	Definition
X (I)	$x_{n-1/2}$ for segment i (i = 1, IX) and $x_{n+1/2}$ for segment IX
XKR (J)	$1/k_r$ for reduced velocity j, j = 1, KVBRW
XN (I)	Control point coordinate, segment i
XMINUS (I)	$X_{n-1/2}$ for segment i
XPLUS (I)	X _{n+1/2} for segment i

The complete symbolic listing is given on the following pages.

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HM150016
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                                     HM150001
                                                      IDELVOL(50), DELTAX(50), XPLUS(50), XMINUS(50), A(50,50), B(50,50),
                                     DIMENSION SVAR(51) SVPLUS(50), SVMINU(50), X(51), XN(50),
AERODYNAMIC INFLUENCE
                                                                                                            DEFINE N2 TAPE TO BE WRITE OUTPUT TAPE NUMBER
DEFINE N3 TAPE TO BE PUNCH TAPE NUMBER
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                   DELVOL(I)=(SVPLUS(I)+SVMINU(I))/2。*DELTAX(I)
                                                                                           DEFINE NI TAPE TO BE READ INPUT TAPE NUMBER
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                           IF(M1-1)109,109,112
WRITE OUTPUT TAPE N2,501,(TITLE(I),1=1,12)
                                                                                                                                                                                                                                                                                                                                                                                                                        N1;131; (XKR(L); L=1, KVBRW)
                                                                                                                                                                                                                                                                                                                                                                                  READ INPUT TAPE N1641, (SVAR(I), I=1, IX1)
                                                                                                                                                                                                                              READ INPUT TAPE N1;1, (TITLE(I), I=1,12)
                                                                                                                                                                                                                                                                                                                          READ INPUT TAPE NI, 40, CBAR, BR, S, SREF,
                                                                                                                                                                                                                                                                    READ INPUT TAPE NI,3, IX, KVBRW, IPUNCH
                                                                         2CH(50,50),C(50,50)(TITLE(12),XKR(50)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                            XN(I)=(XPLUS(I)+XM-FNUS(I))/2.
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                         DELTAX(I)=XPLUS(I)+XMINUS(I)
                                                                                                                                                                                                                                                                                                                                                                FORMAT (4E12.8/(6E12.8))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                 FORMAT(1H1 31X,12A& //)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                SVPLUS(I)=SVAR(N)
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                                                                                                                                                                                                                                                                                                                                             1(X(I), I=1, IX1)
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                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                        KWINUS(I)=X(I)
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                                                                                                                                                                                                                                                                                                                                                                                                                                                                               DO 21 I=1, IX
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                                                                                                                                                                                                                                                                                       FORMAT (1814)
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COEFFICIENTS AERODYNAMIC INFLUENCE

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                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    DF3=(DELTAX(1)+2.*DELTAX(2)+DELTAX(3))*(DELTAX(2)+DELTAX(3))
                                                                                                                                                      /51X,4H BR=1E15.8/52X,3H S=1E15.8/49X,6H SREF=1E15.8//)
                                                       FORMAT(1H1 29X,35H AERODYNAMIC INFLUENCE COEFFICIENTS
                   FORMAT(1H 29X,35H AERODYNAMIC INFLUENCE COEFFICIENTS
                                                                                                                                 47X,114,19H REDUGED VELOCITIES//49X,6H CBAR=1E15.8
                                                                                                                                                                                                                                                                                                            FORMAT(IH 17X, 16H CONTROL PT. NO. 17X, 10H DELTAX(I)
                                                                                                              FORMAT(1H 49X, 11H INPUT DATA//47X, 114, 9H SECTIONS/
                                                                                                                                                                       WRITE OUTPUT TAPE N2,115, (X(I), SVAR(I), I=1, IXI)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                DF2=(DELTAX(1)+DEL⊕AX(2))*(DELTAX(2)+DELTAX(3))
                                                                                            WRITE OUTPUT TAPE N2,4,1X,KVBRW,CBAR,BR,S,SREF
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                      A(1,1)=(2.*SVPLUS(1)*(3.*DELTAX(2)+DELTAX(3))-
                                                                                                                                                                                                                                                                      WRITE OUTPUT TAPE N2,45,((I,DELTAX(I),XN(I)),
                                                                                                                                                                                                                                                                                                                              14X,20H CONTROL ROINT XN(I),//(23X,114,22X,
                                                                                                                                                                                            FORMAT(IH 39X,5H XII),24X,12H LOCAL AREA//
                                                                                                                                                                                                                                                    FORMAT( // 52X,12H OUTPUT DATA //)
                                    25H FROM SLENDER≠BODY THEORY//)
                                                                            25H FROM SLENDER-BODY THEORY//)
                                                                                                                                                                                                               (35X, 1E15, 8, 18X, 1E15, 8))
WRITE OUTPUT TAPE N2,813
                                                                                                                                                                                                                                 WRITE OUTPUT TAPE N2,140
                                                                                                                                                                                                                                                                                                                                                   1E15.8,13X,1E15.8))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                              COEFA=XKR(L)**2/S
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          DO 999 L=1,KVBRW
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                  DO 100 I=1, IX
                                                                                                                                                                                                                                                                                                                                                                                      DO 22 I=1, IX
                                                                                                                                                                                                                                                                                                                                                                                                          22 J=1, IX
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                                                                                                                                                                                                                                                                                                                                                                                                                                                B(I, J)=0.
                                                                                                                                                                                                                                                                                                                                                                                                                                                                   C(I,J)=0.
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                      CONTINUE
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                   KS=1X**2
                                                                                                                                                                                                                                                                                          11=1, IX)
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                                                                                                                                                                                                                                                                                                                                                                      J2=1
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A(I,J)=1./DF7*(-2.*SVPLUS(I)*(DELTAX(I)-DELTAX(N))
                                                                                                                                                                                                                                                                                  2DELTAX(1)+2。*DELTAX(2)+DELTAX(3)}-2。*DELVOL(1)*(2。*DELTAX(1)
                                                                                                                                                                                                                                                                                                                                                                                                                                                      |*DELTAX(1)*(2.*DELTAX(1)*DELTAX(2))*2.*DELVOL(1)*(DELTAX(1)
                                                                                                                                                                                                                                                                                                                                                                    DELTAX(I)*(2°*DELTAX(I)*DELTAX(N)}+2°*DELVOL(I)*(DELTAX(I)
                                                                                                              |+2°+SVMINU(1)*(3°+BELTAX(1)+2°+DELTAX(2)+DELTAX(3)))/DF2
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                           B(I,J)=-1,/DF7*(SVPLUS(I)*DELTAX(I)*DELTAX(N)+SVMINU(I)*
                                                                                  A(1,2)=(2.+SVPLUS(1)+(DELTAX(1)-2.+DELTAX(2)-DELTAX(3))
                                                                                                                                                                                                                                                                                                                                                                                                                          B(1,3)=-1,/DF3*(SVRLUS(1)*DELTAX(1)*DELTAX(2)+SVMINU(1)
                                                        .2. *SVMINU(1) *(4. *DELTAX(1)+3. *DELTAX(2)+DELTAX(3)))/DF1
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                            C(1,1)=1./DF1*(DF1*DELVOL(1)-DELTAX(1)**2/6.(SVPLUS(1)-
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                      SVMINU(1))*(2.*DEL(TAX(1)+3.*DELTAX(2)+DELTAX(3)))*(-1.)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    C(1,2)=-1./DF2+(DE&TAX(1)++2/6.+(SVPLUS(1)-SVMINU(1))+
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                         C(1,3)=+1./DF3*(DELTAX(1)**2/6.*(SVPLUS(1)-SVMINU(1))*
                                                                                                                                                                                                                                                                                                                                                                                            2+2。*DELVOL(1)*(DEL@AX(1)+2。*DELTAX(2)+DELTAX(3)))/DF2
                                                                                                                                                                                                                                                                                                                                        B(1,2)=(SVPLUS(1)*DELTAX(1)*(2.*DELTAX(2)+DELTAX(3))
                                                                                                                                                                                                                                                    IDELTAX(3))-SVMINU(1)*(2.*DELTAX(1)+DELTAX(2))*(2.*
                                                                                                                                        A(1,3)=(-2,*SVPLUS(1))*(DELTAX(1)-DELTAX(2))-2,*
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                             DF8=(DELTAX(M)+DELTAX(I))*(DELTAX(I)+DELTAX(N))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                               C(I,J)=+1,/DF7*(DELTAX(I)**2/6,*(SVPLUS(I)=
                                                                                                                                                                                                                          B(1,1)=(SVPLUS(1)*DELTAX(2)*(2.*DELTAX(2)+
AERODYNAMIC
                                                                                                                                                                    [-2.*SVMINU(I)*(3.*DELTAX(I)+DELTAX(N)))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                              ((DELTAX(1)+2°*DELTAX(2)+DELTAX(3)))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                         SVMINU(I))*(DELTAX#I)+DELTAX(N)))
                                                                                                                                                                                                                                                                                                              3+3 ° *DELTAX(2) +DELTAX(3)) / DF1
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                     ((DELTAX(1)+DELTAX(2)))
                                                                                                                                                                                                IF (XKR(L))25,100,25
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                 IF(J-(M+1))11,11,12
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                   IF (XKR(L))30,97,30
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                     IF (J-M)97,9,10
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                             IF(I-IX)7,8,8
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          DO 97 J=1, IX
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                  2+DELTAX(2)))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                     2+DELTAX(N)))
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AERODYNAMIC INFLUENCE COEFFICIENTS

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                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                      DELTAX(IX) + DELTAX( { X~-1 }) + SVMINU(IX) * DELTAX(IX) * DELTAX(IX-1) +
                                                                                                                                                                                                                                                                                       DF9=(DELTAX(M)+2.*BELTAX(I)+DELTAX(N))*(DELTAX(I)+DELTAX(N))
                          -DELTAX(N)}-2。*SVMfNU(I)*(DELTAX(M)-2。*DELTAX(I)-DELTAX(N)))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                DF6=(DELTAX(IX-2)+2.*DELTAX(IX-1)+DELTAX(IX))*(DELTAX(IX-1)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    DF5=(DELTAX(IX-2)+BELTAX(IX-1))*(DELTAX(IX-1)+DELTAX(IX))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    I*DELTAX(IX-1)+DELTAX(IX-2))+2.*SVMINU(IX)*(DELTAX(IX)-2.*
                                                                                                                .DELTAX(I))-SVMINU(F)*DELTAX(M)*(2.*DELTAX(I)+DELTAX(N))
                                                                                                                                                                                                                                                                                                                 A(I,J)=1./DF9*(-2.*SVPLUS(I)*(DELTAX(M)+3.*DELTAX(I))+
                                                                                                                                                                                                                                                                                                                                                                                                                                    1DELTAX(I))+SVMINU(4)*DELTAX(M)*DELTAX(1)+2.*DELVOL(I)*
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                  DELTAX(IX-1))-2。+SWMINU(IX)*(DELTAX(IX)-DELTAX(IX-1)))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                             C(I,J)=1./DF9*(DELTAX(I)**2/6.*(SVPLUS(I)~SVMINU(I))
A(I,J)=1,/DF8*(2,*SVPLUS(I)*(DELTAX(M)+2,*DELTAX(I)
                                                                                                                                                                                                    I(SYPLUS(I)-SVMINU(#))*(DELTAX(N)-DELTAX(M)))*(-1°)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                            A(IX,IX-1)=1./DF5*12.*SVPLUS(IX)*(3.*DELTAX(IX)+2.
                                                                                                                                                                                                                                                                                                                                                                                                       8(1, 1)=1,/DF9*(SVPLUS(1)*DELTAX(1)*(DELTAX(M)+2.*
                                                                                  B(I,J)=1./DF8*(SVPLUS(I)*DELTAX(N)*(DELTAX(M)+2.*
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                    A(IX,IX-2)=1./DF4*1-2.*SVPLUS(IX)*(3.*DELTAX(IX)+
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                              DF4=(DELTAX(IX-2)+0ELTAX(IX-1))*(DELTAX(IX-2)+2.*
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                         A(IX,IX)=1./DF6*(-2.*SVPLUS(IX)*(4.*DELTAX([X)+
                                                                                                                                                                        C(I,J)=1,/DF8*(DF8*DELVOL(I)+DELTAX(I)**2/6,*
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          B(IX, IX-2)=1,/DF4* SVPLUS(IX)*DELTAX(IX)*(2.*
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                 22.*DELVOL(IX)*(DELTAX(IX)+DELTAX(IX-1)))
                                                                                                                                           2+2。*DELVOL(I)*(DELTAX(N)-DELTAX(M)))
                                                                                                                                                                                                                                                                                                                                                12. *SVMINU(I)*(DELTMX(M)-DELTAX(I))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                        l*(DELTAX(M)+DELTAX(I)) )*(-1.)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                   2DELTAX(IX-1)-DELTAX(IX-2)))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                           | DELTAX(IX-1)+DELTAX(IX)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                 2(DELTAX(M)+DELTAX(E)))
                                                                                                                                                                                                                                                              IF (J-(M+2))13,13,100
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                               IF(XKR(L))32,100,32
                                                                                                                                                                                                                                                                                                                                                                              IF (XKR.(L)) 14,97,14
                                                       IF(XKR(L))31,97,31
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                  2-1)+DELTAX(IX-2)))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                              1+DELTAX(IX))
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                   GO TO 100
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                        HM150153
                                               HM150154
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                                                                                                                                                                                                             1M150161
                        1DELTAX(IX)+2。*DELTAX(IX-1)+DELTAX(IX-2))+SVMINU(I)*DELTAX(IX)
                                            2*(2°*DELTAX(IX-1)+DELTAX([X-2))+2°*DELVOL([X)*(DELTAX([X)+
                                                                                                                 [*(2。*DELTAX(IX)+2。*DELTAX(IX-1)+DELTAX(IX-2))-SVMINU(IX)*
                                                                                                                                        2DELTAX(IX-1)*(2°*DELTAX(IX-1)+DELTAX(IX-2)}+2°*DELVOL(IX}
                                                                                          B(IX, IX)=1./DF6*(SWPLUS(IX)*(2.*DELTAX(IX)*DELTAX(IX-1))
                                                                                                                                                                                                                                                                                                    [*(SVPLUS(IX)~SVMINU(IX))*(2。*DELTAX(IX)+3。*DELTAX(IX-1)
                                                                                                                                                                                                                                                         SVMINU(IX))*(DELTAX(IX)+2.*DELTAX(IX-1)+DELTAX(IX-2)))
                                                                                                                                                                                                                                                                               C(IX,IX)=1./DF6*(D66*DELVOL(IX)*DELTAX(IX)**2/6.
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                           FORMAT(1H 49X,7H 1)/KR=1E15.8//46X,10H NUMBER OF
                                                                                                                                                                                                                                  C(IX,IX-1)=1./DF5*&DELTAX(IX)**2/6.*(SVPLUS(IX)-
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                  TRANSIENT
                                                                                                                                                                                    C(IX,IX-2)=1./DF4**DELTAX(IX)**2/6.*(SVPLUS(IX)
B(IX,IX-1)=-1./DF54(SVPLUS(IX)*DELIAX(IX)*(2.*
                                                                                                                                                             3*(2°*DELTAX(IX)+3°*DELTAX(IX-1)+DELTAX(IX-2)))
                                                                                                                                                                                                           1-SVMINU(IX))*(DELTAX(IX)+DELTAX(IX-1)))*(-1.)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                  17H CONTROL POINTS =113//49X,14H THE
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                               WRITE OUTPUT TAPE N2,814
WRITE OUTPUT TAPE N2,78,XKR(L),IX
                                                                                                                                                                                                                                                                                                                                                                         WRITE OUTPUT TAPE N3,86,XKR(L),J
                                                                    32. *DELTAX(IX-1)+DE&TAX(IX-2)))
                                                                                                                                                                                                                                                                                                                                                                                                                                            WRITE OUTPUT TAPE N3,34,1X,J1
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                        C(I,J)=COEFZ/CBAR*O(I,J)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                            A(I,J)=COEFZ*CBAR*A(I,J)
                                                                                                                                                                                                                                                                                                                                                                                              FORMAT (1E12.8,65X,113)
                                                                                                                                                                                                                                                                                                                             2+DELTAX(IX-2)))*(-1.)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                     FORMAT(114,73X,113)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                             B(I,J)=COEFZ*B(I,J)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                        IF (XKR (L))46,46,71
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                   IF (XKR(L))80,70,80
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                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                 COEFZ=2,/SREF
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                              DO 47 [=1, IX
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                     DO 47 J=1, IX
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          00 81 I=1, IX
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                 J=1, IX
                                                                                                                                                                                                                                                                                                                                                       CONTINUE
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                       CONTINUE
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                             J1=J1+1
                                                                                                                                                                                                                                                                                                                                                                                                                          J1=J1+1
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AERODYNAMIC INFLUENCE COEFFICIENTS

501 501 501 501 501	HM150197 HM150198 HM150199 HM150200 HM150201	HM150203 HM150204 HM150205 HM150206	HM150208 HM150209 HM150210 HM150211 HM150213	HM150214 HM150215 HM150216 HM150217 HM150219 HM150219	5022 5022 5022 5022 5022 5022
2 20 01 34	GD TD 702 833 J1=J2+1 702 J2=J1+(IX+IX/22)*3±4 302 WRITE OUTPUT TAPE N2,66 66 FORMAT(// 49X,15H THE CHD MATRIX /) CALL MPRINT(8,1X,50,N2)	02	WRITE OUTPUT TAPE WRITE OUTPUT TAPE 2 FORMAT(1H 49%,7H 1 1 17H CONTROL POIN CALL MPRINT(A,IX,I IF(IPUNCH)999,845,	145 IF(J1-2)847,846,843 146 J1=J2 6D TD 888 147 J1=J2+1 188 J2=J1+IX+IX/22*IX+2 WRITE DUTPUT TAPE N2,229,J1,J2	3 CHS=606060233062 CALL MRUNCH(A, IX, IX, 0, 0, 1, CHS, 50, N3) IF(XKR(L))79,999,79 79 CONTINUE B CHD=606060233024 B CHI=606060233031 CALL MPUNCH(B, IX, IX, 0, 0, 1, CHD, 50, N3)
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HM150250
                                                  HM150229
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                                                                                                                                                                                                        FORMAT(1H 12X,4HREAL 11X,9HIMAGINARY 11X,4HREAL 10X,
                                                                                                                                                       16H CONTROL POINTS=113//48X,16H THE OSCILLATORY
                                                                                                                                      FORMAT(1H 49X,7H 11/KR=1E15.8//46X,10H NUMBER OF
                                                                                                                                                                                                                          9HIMAGINARY 10X, SHREAL 11X, 9HIMAGINARY //)
                                                                                                                                                                                                                                                                                                                CH(I,J)=COEFA*A(I,KZ)-(1,/(BR**2*S)*C(I,KZ))
                                                  FORMAT(/38X,24H PUNCHED CARDS NOS. HMI5113,
AERODYNAMIC
                                                                                                                                                                                                                                                                                                                                                                                                                                     CALL MPUNCH(CH, IX, EX2, 0, 0, 1, CHM, 50, N3)
                                  CALL MPUNCH(C, IX, IX, 0, 0, 1, CHI, 50, N3)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                END(1,1,0,0,0,0,0,0,1,0,1,0,0,0,0)
                                                                                                   WRITE OUTPUT TAPE N2,814
WRITE OUTPUT TAPE N2,85,XKR(L),IX
                                                                                                                                                                                                                                                                                                                               CH(I, J+1)=XKR(L)/(BR*S)*B(I,KZ)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          WRITE DUTPUT TAPE N2,229,J1,J2
                                                                                                                                                                                                                                                                                                                                                                 CALL MPRINT(CH, IX, 1X2, 50, N2)
                                                                                                                                                                                       WRITE GUTPUT TAPE N2,919
                                                                                                                                                                                                                                                                                                                                                                                  [F(IPUNCH)999,65,999
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                        J2=J1+2+(IX+IX/22)#2
                                                                                                                                                                                                                                                                                                                                                                                                                                                     IF(J1-2)887,889,887
                                                                   1 10H THRU HM15113)
                                                                                                                                                                                                                                                                                                                                                                                                                     CHM=606060233044
                                                                                                                                                                                                                                                                             DO 35 J=1, IX2, 2
                                                                                                                                                                                                                                                             DO 35 I=1, IX
                                                                                                                                                                         5H CASE//)
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                           END FILE N3
                                                                                      60 TO 999
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                        GO TO 631
                                                                                                                                                                                                                                            IX2=2*IX
                                                                                                                                                                                                                                                                                                                                                  CONTINUE
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                            CONTINUE
                                                                                                                                                                                                                                                                                              KZ = J/2 + 1
                                                                                                                                                                                                                                                                                                                                                                                                   CONTINUE
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                           JJ = J2 + 1
                                                                                                                                                                                                                                                                                                                                                                                                                                                                        11=12
                                                   229
                                                                                                                                                                                                           919
                                                                                                                                                                                                                                                                                                                                                                                                                                                                        889
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                            666
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          887
                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                                          631
                                                                                                                                        85
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10/11/62

-		N AND EQUIVALENCE STATEMENTS		SVMINU 12366 30116 XMINIS 12065 27441		N, OR EQUIVALENCE STATEMENTS	DEC OCT	1950	1946	DF4 1942 03626	1938	1934	1930	1926	1922	1918	FORMAT STATEMENTS	EFN LOC	12 34 033	22 66	2L 85	0	PE 814	
BY PROGRAM		N DIMENSION				DIMENSION,	00.1	. 03637	03633		03623	03617	03613	03607	03603	03577	PROGRAM	707				03540		
USED		NI 9NI	DEC 6953	12467	12316	COMMON	DEC	1951	1947	1943	1939	1935	1931	1927	1923	1919	SOURCE	EFN	4	45	78	131	813	
STORAGE NOT		LES APPEARING	3	SVAR	×	APPEARING IN CO		CHD	COEFA	0F3	DF7		5	KVBRM	Z	SREF	FOR		8)4	8) 1D	8) 2E	8)43	8) PD	
v	0CT 77461	R VARIABLES	0CT 22355			NOT APPE		0	0		0	0	0	0	0	0	AND LOCATIONS	707	03546	35	33	4	35	32
	DEC 32561	INS FOR	DEC 9453	12215 12015	12115	LES	DEC	1952	1948	1944	1940	1936	1932	1928	1924	1920	01.5	EFN	G	41	72	115	501	919
		STORAGE LOCATION	80	DELVOL	XPLUS	IS FOR VARIAB		CBAR	CHS	0F2		FONCH	XI	XX:	Σ	Z	SYMB		$\overline{}$	_	2	8)37	u.	S
	OCT 30264	STOR	OCT 27261	27605 30200	27751	LOCATIONS	၁	364	363	03631	362	362	361	561	360	360		\Box	03547	35	33	m (32	8
	DEC 12468		DEC 1953	165 416	265	STORAGE L	EC	95	94	1945	94	63	93	92	92	92		EFN	~	40	29	86	229	2
			∢	DELTAX SVPLUS	×	STO		BR	CHA	DEI	DE5	DF9	IX2	: : د	I				811	-	CO .	8)2M	-	22

10/11/62

			LOCATIONS	
PROGRAM	DEC OCT 2767 77777 1906 03562 1910 03566 1914 03572 656 01220 262 00406 15 00017 1473 02701 647 01207	DEC 0CT 5 00005 4 00004 RY	(STH) S AND OCTAL	IFN LOC 87 00350 116 01043 125 01212 133 01371 147 01665 166 02236 180 02334
IN SOURCE PR	4) 3)60)65 200 112 200 401 526	VECTOR (FIL) (STH) FROM LIBRARY	(RWT) FORMULA NUMBERS	EFN 112 6 10 13 32 80 834
IT APPEARING	642 642 642 642 643 9414 1114 113 546 546 546	IN TRANSFER DEC OCT 8 00010 3 00003 ES NOT OUTPUT	(RTN) INTERNAL FOR	FN LOC 66 00224 109 00606 122 01134 132 01355 140 01525 164 02222 179 02326
SYMBOLS NOT	3) A)103 C)64 C)103 C)900 D)11C D)32E D)51C	OF NAMES (EFT) (RWT) SUBROUTIN	(FPT) CORRESPONDING	EFN 1 109 25 30 12 8 47 301
IONS FOR OTHER	DEC DCT 1634 03142 1608 03110 1908 03564 1912 03570 1916 03574 844 01514 843 01513 1474 02702 842 01512	LOCATIONS DEC OCT 7 00007 2 00002 ENTRY POINTS TO	(FIL) ERS WITH	1FN LDC 63 00213 102 00452 119 01070 129 01265 138 01515 162 02207 172 02270 183 02342
LOCATION	2)	MPUNCH (RTN)	(EFT) FORMULA NUMB	EFN 21 21 31 46 46 702
	DEC 0CT 1896 03550 1652 03164 1907 03563 1911 03567 1915 03573 760 01370 1597 03075 559 01057 14 00016	DEC OCT 6 00006 0 00000 1 00001	MPUNCH EXTERNAL	IFN LDC 25 00020 93 00373 117 01047 126 01221 136 01434 153 02133 170 02260 182 02337
	(1) (2)(6) (2)(6) (2)(10) (2)(2)(11)(6) (3)(4)(11)(6) (4)(11)(6)(11)(6) (5)(11)(6)(11)(6) (6)(11)(6)(11)(6) (7)(11)(6)(11)(6) (7)(11)(6)(11)(6) (7)(11)(6)(11)(6) (7)(11)(6)(11)(6) (7)(11)(6)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(11)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)(6) (7)	MPRINT (FPT) (TSH)	MPRINT (TSH)	EFN 2 22 22 7 110 1100 8133

1.

	AERODYNAMIC	INFLUENCE COEFFICIENTS 10/11/62	PAGE 11
0000 -342647633460 00 (FPT		00045 +000000003546 010	PZE 8)3
0001 -346362303460 00 (TSH	CD 1 (-1000000000000	STR
00002 -345163453460 00 JRTN)	BCD 1(RTN)	0000003614 0	STQ 1X
0003 -345166633460 00 4RWT	00	-100000000000	STR
0004 -346263303460 00 JSTH		0 109800000090-	STO KVBRW
0005 -342631433460 00 . (FIL	CD 1(F	-100000000000	STR
0006 -044751314563 00 NPRI	CD IMPRI	-060000003620 -01	IPUN
0007 -044764452330 00 MPUNC	CD IMPUN	+007400400002 01	(RIN)
0010 -342526633460 00 &EFT	CD 1	+007400403110 01	A)10
0011 +050000000000 00 \$\$	LA 2	-053400203570 01	C)102,
0012 +060100003171 01	10	-053400403614 01	
0013 +05000000000000000	LA (FP	-063400403566 01	5()
00 0100000000000 + 000		+050000003614 01	
0015 +060000077462 01	TZ 4)-2	+040000003145 01	ADD 21+3
0016 -053400203570 010 016	XD C)102	+060100003616 01	X
0017 -053400403566 010 B)40	XD C165	-050000003603 01	
0020 +050000003142 010 25		-063400403170 01	9
0021 +060100003603 0	10	+007400400001 0	(TS
0022 +050000003143 01	LA 2	+000000003545 01	PZE 8)18
0023 +060100003602 01	TO N	-053400403170 01	4 (9
0024 +050000003144 01	ΓA	-100000000000 0	STR
0025 +060100003601 01	T0 N	-060000003640 0	STO CEAR
0026 +050000003145 01	LA 2	-100000000000000	STR
0027 +060100003605 01		-060000003641 0	STQ BR
0030 -050000003603 01		-100000000000 00	STR
0031 -063400403170 01	(9 QX	-060000003576 0	STO S
0032 +007400400001 01		-100000000000 0	STR
0033 +0000000003547 01		-060000003577 01	S
0034 -053400403170 01	+(9 0x	-053400103145 0	2)
0035 -053400103145 010 3		+050000003616 01	CLA IX1
18 00 000000000001- 9800		+062200000107 01	40
0037 -060000127360 01		-100000000000	STR
0040 +1000011000041 010 31A		-060000130035 01	
10 9600014100036 01		+100001100107 010 40	
833	SS	-300000100104 010 40A	
0042 +007400400002 01		-063400403170 010 42	(9
43 -0500		+007400400002 01	TSX (RTN) 4
0044 +007400400001 01	SX (-053400403170 01	(9 Q

	AERODYNAMIC	DEFFICIENTS 1	
43A	CAL N1 SXD 6)+4,4	01000036	STO I CLA I
	SX (TSH)	+040000003145 010	
	2E 8	+060100003600 01	
	X Q X	-053400203600 01	
44V	XD 2)+3,	-063400203564 010	
	LA IXI	+050000230035 01	X+1,2
4	10 45 TD	+05010012/524 01	
	TO < V	+060100127442 010	
⋖	XI *+1°	+050000127524 01	XPLUS+
5A2	XL 45A	+030200127442 010	XMINUS
	y OX	+060100127606 01	DEL TAX+1
	SX	+050000127524 01	XPLUS+1
	9 QX	+030000127442 01	XMINUS
8A	AL N	+024100003152 01	
	7+(9 QX	-060000127752 01	
) XS	+050000230264 01	SVAR+1,
	8 3Z	+060100130201 01	SVPLUS+
	9 0x	+050000130264 01	SVAR+1,
⋖	Š	+060100130117 010	
	LA KVBR	+050000130201 01	SVPLUS+
	10 5	+030000130117 01	NAINO+
⋖	TR	+024100003152 01	3)
	TO XK	+026000127606 01	DEL TAX+
7	* IX	+060100127670 010	DELVOL+1,
A2	XL 504	63A	
⋖	9 0X	213 +100001100214 010 63	+
	SX (RTN)	214 -063400103617 010	-
	9 QX	215 -300000100162 01	55A
⋖	AL N	216 +050000003146 010 64	2)
	XD 6)+4	217 +060100003613 01	7
	X	220 +05000003605 01	Σ
	XD 6)+4,	221 +040200003145 01	2)
4	xD 2)+3	222 +010000000224 01	66A
	LA IX	0	87
	TD 63	224 -050000003602 01	Z
	QX	225 -063400403170 01	9

				•	AERODYNAMIC	INFLUENCE CO	COEFFICIENTS	10/11/62	PAGE 13
00226	4004000	010		TSX P7F	(STH),4	00274	-063400403170 +0074004000004	010	SXD 6)+4,4
023	05340040317			×	6)+4°4	10	, W		833.1
023	05340010314	-	~		3	2	-053400403170	010	6144
023	05600012736	-	€8A	۵	7	3	4		21+3
023	0000000000	00		F		M	-	,	ĭ X ĭ
023	10000110023	_	Ø	×	*+1,11,1	3	-	_	92
023	0001410023	$\overline{}$	68A2	×	68A,1,12	3	3	-	X+1
023	6340040317		0	8	6)+404	3	0	0	STR
023	00740040000	010		SX	(FIL),4	M	9	011	LOQ SVAR+1,1
024	05340040317	_		2	6)+484	3	0	00	STR
024	5000000360		FIA	A	N2	3	-	10 7	*
024	06340040317	$\overline{}$		2	6)+404	3	Ó	10 76A	16
024	00740040000	7		S	(STH),4	3	~	10 7	6)
024	00000000353	_		7	8)PD	3	0	~	(FIL)
024	0740040000	~		S	(FIL),4	3	~		•
024	05340040317			×	6)+404	3	0	_	Z
024	05000000360	_	72A	⋖	N2	M	~	_	4+(9
025	6340040317			×	61+4,4	3	0	-	-
025	00140040000	-		S	(STH) 94	3	_	-	ω
025	000000000347	~		7	8)4	3	0	010	(FIL)
025	05340040317			×	6)+404	3	7	-	7+(9
025	5600000361	10	73A	ā	×Ι	3	0	_	
025	10000000000	0		<u>-</u>		3	\vdash	-	9
025	02600000360	010		۵	KVBRW	3	0	-	ST
025	000000000	0		F		3	•		
026	05600000364	010		ã	CBAR	3	~	-	4 (9
026	10000000000	0		<u>-</u>		3	40010314	-4	5)+
026	5600000364	010		Õ	B R	3	5000000361	$\overline{}$	X
026	10000000000	0		-		M	220000034		84
026	05600000357	010		۵	S	(1)	540010000	00	PXD 0,1
026	0000000000	0		F		3	010000361	010	STO 1
026	05600000357	010		ā	SREF			82A	BSS
026	000000000	00		=		33	5600000361	10 84	rod I
027	06340040317	-		×		33	0000000000	0	
027	7 40040000	010		S	(FIL) o4	00336	+056000127606	010	LDQ DELTAX+1,1
027	5340040317	***		$\overline{\times}$	614494	33	10000000000	0	
027	500000360	بس	¥4.A	₹	N2	34	5600012775		

	AERODYNAMI C	INFLUENCE COEFFICIENTS 10/11/62	PAGE 14
0341 -100000000000000000	STR	03102 01	244A
0342 +100001100343 01	×	06 -053400203570 010 0)	
00343 -063400103617 010	SXD I,1	+056000127344 010 9	XKR+
0344 -300000100334 010 84	×	0 +026000127344 01	
0345 -063400403170 01	×	0411 +024100003576 01	FDP S
0346 +007400400005 01	S	-060000003633	ပ
0347 -053400403170 010	ΩX	3 +056000003614	
0350 +050000003145 01		+02000003614	H
0351 +060100003612 01	-	00415 +076700000021 00	ALS 17
0352 -053400103145 01	OX	+060100003610	¥
0353 +050000003614 01	LA I	-053400103145	
0354 +062200000377 01	-	-063400103567	SXD C)100,1
0355 +056000003614 01	1 00	+056000003614	-
0356 +020000003147 01	PY 2	00422 +020000003147 010	MPY 21+5
0357 +076700000021 00	LS 1	+07670000021	-
0360 +062200000374 01	TD 9	+062200001522	138
0361 +056000003614 0	1 00	-053400103145	7
0362 +020000003147 01	۵.	-063400103562	ບ ຶ ບ
0363 +076700000021 00	_	+050000003614	CLA IX
0364 +062200000375 01	\vdash	+062200002141	-
	S	-075400100000	
0365 +050000003153 010 9		+060100003617	
00366 +060100127262 010	2	+050000003617	_
0367 +050000003153 01	LA	+040200003145	
0370 +060100122356 01	10	+060100003604	STO M
0371 +050000003153 01	LA	-053400203604	Σ
0372 +060100110546 010	\vdash	-063400203565 010	Ĉ
93	S	+0500000030+	-
0373 +100062100374 010 93A	TXI *+1,1,50	0441 +040000003145	
0374 -300000100365 010 93	XL 90	0442 +060100003600	
0375 +200000100376 010 93A	+* XI	43 -053400203600	N , 2
0376 +100001100377 010	×	44 -063400203564	Ĉ
0377 -300000100365 01	×	0445 +050000003617	
0400 +050000003142 010 94	LA 2	0446 +040200003145 01	21+
	10 M1	0447 +010000000452 0	E 10
0402 -053400103145 01	XD 2 3+	0450 +012000001043 01	L 116
0403 -063400103563 01	XD C \\ 0.1	451 +002000001043 010	A 116
0404 +050000003607 01	LA KVB	0452 +056000003152 01	LDQ 3)

	AERODYNAMI	IC INFLUENCE COEFFICIENTS 10/11/62	PAGE 15
0453 +026000027604 01	MP DELT	521 +060100003553 01	0 13+
0454 +030000027605 01	AD D	522 +050000003551 0	CLA 11+1
0455 +030000027603 01	AD DELT	00523 +030000027603 01	E
0456 +060100003551 01	TO 1	524 +013100000000 0	XCA
0457 +050000027605 01	LAD	525 +02600003152	FAP 37
0460 +030000027504 01	AD DELTA	00526 +013100000000	XCA
00 000000000000000000000000000000000000	ပ	527 +026000030200 01	SV
0462 +026000003551 01	MP 1	530 +030200003553 01	4 1 2
0463 +060100003631 01	10	531 +024100003631 01	DF
0464 +050000027604 01	LA DELTAX-	00532 -060000027261 01	
0465 +030000027603 01	AD DELTA	00533 +056000003152 01	ሊነ
10 15500001090+ 9940	TO 1)+	534 +026000027604	DEL
0467 +050000027605 01	LA DEL	535 +060100003551 01	-4
0470 +030000027604 01	AD DELTA	536 +056000003154 01	3)+
0471 +013100000000 00	ပ	537 +026000027605 01	Ω
0472 +026000003551 01	MP 1	540 +030000003551 01	*
0473 +060100003630 01	TO D	541 +030000027603 01	۵
0474 +050000027604 01	LA DEL	00542 +013100000000	XCA
0475 +030000027603 01	AD DELTAX-	00543 +026000003152	FMP 33
0476 +060100003551 01	TO 1)+	544 +013100000000	XCA
0477 +056000003152 01	0	545 +02600030116 01	SVM
0500 +026000027604 01	MP DELTA	00546 +060100003553 01	+
0501 +030000027605 01	ΔD	547 +050000027605	
0502 +030000027603 01	AD DELTA	00550 +030200003551 01	نام. بستن سنند
0503 +013100000000 00	\circ	551 +030200027603 01	FSB DELTAX-2
0504 +026000003551 01	MP I)	552 +013100000000	XCA
0505 +060100003627 010	-	553 +02600003152	FMP 33
0506 +056000003154 01	00 31+	554 +0131000000000	XCA
0503 +026000027604 01	MP DEL	555 +026000030200 01	SVP
0510 +060100003551 01	10 13+	556 +030000003553 01	~
0511 +056000003155 01	00 3)	557 +024100003630	
0512 +026000027605 01	MP DEL	560 -060000027177 01	A-5
0513 +030000003551 01	AD 134	561 +056000003154 01	m
0514 +030000027603 01	A D	00562 +026000027605 01	DELT
0515 +013100000000 00	ပ	563 +030000027604 01	D DELTA
00516 +026000003152 010	FMP 33	64 +01310000	XCA
0517 +0131000000000 00	Ç	565 +026000003152	Σ.
0520 +026000030116 0	Σ	266 +0131000000000	(J

PAGE 16	SVM	DE SV	0 1 0 8	LDQ 3) FMP DELTAX-1 STO 1)+1 FAD DELTAX FAD DELTAX	2 0	FMP DELTAX FAD 13+1 FAD DELTAX-2 XCA FMP SVMINU	FMP DELTAX STO 1)+5 CLA 1)+1 FAD DELTAX=2 XCA FMP SVPLUS
10/11/62	010 000 010 010	000	010 010 010	010 110A 010 010 010	000 010 010 010	010 010 00 010	010 010 010 010 010
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INFLUENCE COEFFICIENT 00751 +03000002 00752 +06010000 00753 +05000003 00754 +03020003
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'62 PAGE 18	SXD C)100	C 1 2 0 2 s	A LDQ 3)	LXD C)60,1	FMP DELTAX+1,1	LXD C)64,2	FAD DELTAX+1,2	C)62,4	DELT	1)+1	CLA DELTAX+1,2	DELTAX+1,				100 3	a	DELT		FMP 3)	XCA	FMP SVMINU+1,1	+ / [DELT	DEL		FMP 3)	XCA	FMP SVPLUS+1,1	CHS		۵	3	C 10	STO A+1,2	C)61	XKR &]
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146 +0131000000000	00	CA	+075000000000	CHS
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151 +026000127606	010	d X	+012000001355	
152 +060100003554	010	101	-053400103562	
153 +056000427606	010	DO DELTAX+1,	+050000127606 010	
154 +026000130201	010	랖	-053400203564	
155 +013100000000	00	ت	+030000227606	FAD DELTAX+1,2
156 +026000127606	010	E D	+060100003551	
57 +030000003554	010	AD 1)+	-053430403565	
160 +030000003552	010	AD 1	+050000427606	DE
161 +024100003623	010	o do	+030000127606	DEL TAX+1
162 +026000003156	010	Ī	+013100000000	XCA
163 +076000000002	00	HS	+026000003551	FMP 11+1
164 -053400203567	010	2	01232 +060100003622 010	
165 +060100222356	010	10	+056000003152	
166 +050000127606	010 A23A	LA	+026000127606	FMP DELTAX+1,1
+030000427606	010	D DELTAX+1,	+060100003551	STO 11+1
170 +060100003551	010	1 0	+07.60000000000	CHS
171 +050000130201	010	LA	01237 +030000427606 010	DELTAX+1,
172 +030200130117	010	SB SVMINU+1,	+030200227606 0	FSB DELTAX+1,2
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PAGE 21		8 AX+ 1		XCA FMD 3)	1	FMP DELVOL+1,1			SVMINU+ I s	FMP DELTAX*1,4	11+3		DEL TAX+	DELTAX+1,		FMP SVPLUS+101		DEL	FAD 11+3	+		3)+4	LXD C1100,2			,	DEL TAYAR	1)+2	SVPLUS+1,	SVM INU+1	1)+3	DELTA	FMP DELTAX+1,1
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PAGE 22	FMP SVMINU+1,4 STO 1)+2	3	DELTAX+1,			FXP US		FMP SVPLUS+1,4		,=0	FDP DF4	FMP 33+4	LXD C)102,2	A+101,2	CLA DELTAX+2,4	DELTAX+1,	STO 13+1	DELTAX+3	DEL TAX+2,	XCA	p-4	u	ñ	FMP DELTAX+2,4	tured.		DEL TAX+1	DELTAX+3,	XCA	FMP 3)	⋖	٠,	~ O	or Or	۰	13+1	FAD DELTAX+3,4
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	02042	204	204	204	204	202	205	205	205	205	205	205	205	206	206	205	200	206	206	206	206	207	207	207	207	207	207	207	207	210	210	210	210	210	210	210	210

	AERODYNAMIC	INFLUENCE COEFFICIENTS 10/11/62	PAGE 26
2155 +007400400005 0	×	62 010 164A LDQ	A+1,1
2156 -053400403170 010	S	2223 +02600003632 010 FMP	Ē
2157 +050000003613 01	LA	2224 +013100000000 00 XCA	
2160 +040000003145 01	00	2225 +02600003640 010 FMP	CBAR
2161 +060100003613 010	0	2226 +060100127262 010 ST0	1,1
2162 -050000003601 01	AL	2227 +100062102230 010 164A1 TXI	1
2163 -063400403170 01	+(9 QX	2230 -300000102222 010 164A2 TXL	164A,1
2164 +007400400004 01	SX (ST	2231 +200000102232 010 164A3 TIX	*+101
2165 +000000003354 01	3 2	2232 +100001102233 010 TXI	-
2166 -053400403170 01	+(9 QX	2233 -300000102222 010 164A4 TXL	A , 1
2167 +056000003614 01	0	2234 +050000227344 010 165A CLA	7+
2170 -100000000000 00	-	2235 +010000002435 010 165A1 TZE	193A
2171 +056000003613 0		2236 -053400103145 010 166A LXD	2)+3,1
2172 -100000000000 00	\vdash	2237 +050000003614 010 CLA	
2173 -063400403170 01	×	2240 +062200002264 010 STD	170A4
2174 +007400400005 01	S	2241 +056000003614 010 LDQ	IX
2175 -053400403170 01	×	2242 +020000003147 010 MPY	21+5
2176 +050000003613 01	LA	2243 +076700000021 00 ALS	17
2177 +040000003145 01	00	2244 +062200002261 010 STD	170A2
2200 +060100003613 01	10	2245 +056000003614 010 LDQ	XI
2201 +050000003152 0	LA	2246 +020000003147 010 MPY	21+5
2202 +024100003577 01	оb	2247 +076700000021 00 ALS	17
2203 -060000003632 01	TQ COE	2250 +062200002262 010 STD	170A3
2204 +050000227344 01	LA	167A	
2205 +0100000002207 010 161	ZE 162	2251 +050000003632 010 168A CLA	w
2206 +012000002621 01	PL 219	2252 +024100003640 010 FDP	CBAR
2207 -053400103145 01	XD 2)+		
2210 +050000003614 01	LA	2254 +060100110546 010 STO	_
2211 +062200002233 01	10	2255 +056000003632 010 169A LDQ	EF
2212 +056000003614 01	00	2256 +026000122356 010 FMP	8+1,1
2213 +020000003147 01	γ	2257 +060100122356 010 STO	
2214 +076700000021 00	rs	170A BSS	
2215 +062200002230 01	TD	2260 +100062102261 010 170A1 TXI	-
2216 +056000003614 0	000	2261 -300000102251 010 170A2 TXL	_
2217 +020000003147 01	ΡY	2262 +200000102263 010 170A3 TIX	*+1,1
220 +07670000002	_	2263 +100001102264 010 TXI	
2221 +062200002231 0	TD	A4 TXL	167A,1
	S	2265 +050000227344 010 171A CLA	7

PAGE 27		CLA J2	7	TRA 183A	CLA J2	7		-	G	7	CLM	-		CLA IX			7			ADD 2)+7		1844	c)65	N2	S	æ	<u>.</u>			മ	H	H	TSX 2)+5	Z	A) 10	ပ	١X،	X
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-64	02266 +010000002435 01	2267 +012000002621 01	2270 -050000003602 01	2271 +007400400004 01	2272 +0000000003513 01	2273 +007400400005 01	02274 -050000003602 010 173A	2275 +007400400004 01	2276 +000000003351 01	2277 +056000227344 01	2300 -100000000000 0	2301 +056000003614 0	2302 -1000000000000 0	2303 +007400400005 01	2304 -050000003602 01	2305 +007400400004 01	2306 +000000003322 0	2307 +007400400005 01	376	7E 010 900000000100+ 0152	2311 +007400027261 01	2312 +007400003614 01	2313 +007400003614 01	2314 +007400003147 01	02315 +007400003602 010	2316 +007400403110 01	2317 -053400103570 01	2320 -053400403614 01	2321 -053400403614 01	2322 +050000003620 010 37	2323 +0100000003326 010 178	2324 +012000002361 01	2325 +002000002361 01	2326 +050000003613 01	2327 +040200003142 01	2330 +010000002333 01	2331 +012000002337 01	2332 +00200002337 01

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	AERODYNAMIC	INFLUENCE COEFFICIENTS 10/11/62	PAGE 29
2513 +040000003613 01	00	556 +060200003637 01	
2514 +040000003142 01	0	62 0	M
02515 +060100003612 010	10	560 +060200003636 010	SLW CHI
2516 -050000003602 01	AL N	214A	
2517 -063400403170 01	9 QX	+007400400007 010 215	
2520 +007400400004 01	S	+007400022355 01	TSX B
2521 +000000003257 01	2E 8	3614 01	Ħ
2522 -053400403170 01	×	+007400003614 0	
2523 +056000003613 01		+007400003146 01	2)+
2524 -100000000000 00	-	+007400003146 0	5
2525 +056000003612 0	0	+007400003145 01	2)+
2526 -100000000000 00	-	+007400003637 01	ပ
2527 -063400403170 01	×	+007400003147 01	TSX 2)+5
2530 +007400400005 0	S	+007400003601 01	Z
2531 -053400403170 01	×	+007400403110 01	A)10
206A	S	-053400103570 01	C) I
2532 -050000003160 010 2	⋖	-053400403614 01	ΙΧŷ
010	I	-053400403614 010	—
3 08A	S	-063400403170 010 2	6)+4
2534 +007400400007 010 209	S	+007400400007 010 2I7	TSX MPUNCH,4
2535 +007400027261 01	X	+007400010545 01	ں
2536 +007400003614 01	SX I	+007400003614 01	⊷
2537 +007400003614 01	SX I	+007400003614 01	
2540 +007400003146 01	SX 2	+007400003146 01	2)+
2541 +007400003146 01	SX 2)+	+007400003146 01	7
2542 +007400003145 01	SX	+007400003145 01	2)+
2543 +007400003634 01	S	+007400003636 01	CHI
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2545 +007400003601 01	S	+007400003601 01	TSX N3
2546 +007400403110 01	SX A)10	+007400403110 01	A 3 10
2547 -053400103570 01	XD C	-053400103570 01	C)1
2550 -053400403614 01	Q X	-053400403170 -01	6)+48
2551 -053400403614 01	XD I	-053400403614 01	IΧ
2552 -063400403566 01	Š	-053400403614 01	×ř
2553 +050000227344 010 210	LA XK	-063400403566 01	C)65
2554 +010000003074 01	132	+002000003074 010 21	0)3
211	SS	621 -050000003602 010 219	AL NZ
12	⋖	07922 +00/400400004 010	TSX (SIH) 84

	AERODYNAMIC	INFLUENCE COEFFICIENTS 10/11/62	PAGE 30
+000000000513 01	ZE 8)	2671 -053400403553 01	11+3
+007400400005 010	TSX (FIL) p4	-063400403571 0	03
-050000003602 01	AL N2	2673 -053400403145 01	2)+31
+00740040004 01	SX (S	2674 -063400403573 01	C) 2
+000000003242 01	ZE 8)2	2675 +050000003614 01	
+056000227344 01	DQ XK	2676 +062200002770 01	7
-100000000000 00	-	2677 -075400400000 0	ő
+056000003614 0	\Box	2700 +060100003617 010	I
-100000000000 00	-	2701 -053400103563 010 015	C
+00740040005 01) XS	2702 -053400403567 010 D142	C)100
-050000003602 0	AL N2	2703 -053400203145 010 225	21+3,2
+00740040004 01	S) XS	2704 -063400203572 01	C) 20
+000000003213 01	2E 8)	2705 +050000003615 01	Χĩ
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+056000003142 01	00 2)	2707 -075400200000 00	ô
+020000003614 01	PΥΙ	2710 +060100003611 010	
+076700000021 00	LS 1	2711 +050000003611 01	
+060100003615 01	TO IX	2712 +076500000043 00	
-053400103145 01	XD 2)+3	2713 +022100003142 0	7
-063400103567 0	XD C)1	2714 +076000000000 0	
+050000003615 01	LA IX2	2715 +076300000022 0	LLS 18
+040200003145 01	UB 2)	2716 +013100000000 0	
+040000003142 01	DD 2	2717 +040000003145 01	7
+076500000043 00	RS 3	2720 +060100003606 01	¥
+022100003142 0	VP 2	2721 -063400403170 0	44 (9
+020000003142 01	PY 2	2722 +007400403125 01	A)103
+076500000022 00	RS 1	2723 -053400203571 01	C)103,
+020000003147 0	PY 2	2724 -053400403170 010	6)+40
+076300000043 00	LS 3	2725 +056000003641 01	Ω
+062200002757 01	TD 229	2726 +026000003641 01	ω
+050000003167 0	LA 6	2727 +0131000000000 0	XCA
+060100003553 01	TO 1)+	2730 +026000003576 01	S
+056000003606 01	D0 X	2731 +060100003552 01	17
+020000003147 01	PY 2	2732 +050000003156 01	33+
+076700000021 0	LS 1	2733 +024100003552 01	1 3 +
+040200003147 01	\rightarrow	2734 +026000210546	FMP C+1,2
+040000003553 0	DD 1)+	2735 +060100003553 01	+ (
+060100003553 01	TO 114	2736 +056000003633 01	COE

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02737	+026000227262 010	0	d ₩ L	A+1,2	03003	+010000003006	010 232A1	TZE 233A
4/7	030200003553 01	.	20	11+3	200	10500000710	٠,	٦ . ا ل
274	060100415452 01	1	10	CH+1,4	300	020000307	10	E) 12
274	056000003641 01	D 228A	O C	BR				
274	026000003576 01	c	J.	S	90020	-050000003163	34	3
274	0601000003551 01	C	10	1)+1	300	0602000363		
274	50000127344 01	C	LA	XKR+1,1			235A	BSS
274	024100003551 01	C	DP	1)+1	301	40040000	10 236	TSX MPUNCH,
274	26000222356 01	C	ďΣ	B+1,2	301	0740001545	-	
275	060100415370 01	C	10	H-4	301	1361	ri	-
		29	S		301	40000361	_	×
275	100144402752 01	0 229Al	X	+1948	301	3314	pared;	2)+
275	53400203572 01	C	Q	12000	301	+100000		23
275	100002202754 01	C	×	\$2	301	40000314		2 } +
275	063400203572 01	C	Š	2003	301	+00000+		ں
275	063400203611 01	0	QX	J ₉ 2	302	+00000+	 4	2)
275	00000202711 01	229	×	6 A	302	10000360	-	Z
275	00000402760 01	229 A	×I	-	302	+0040311	1	A) 102,
276	100001402761 01	C	×ĭ	1,40	302	+0020357	$\overline{}$	S
276	53400203571 01	0	X	103,	302	19604004		IXo
02762	100001202763 01	0	ΧI	*+1,201	03025	-063400403566	010	C) G
276	63400203571 01	C	2	103,	302	00000361	~~	٦
276	053400203573 01	C	Q Q	201,	302	20000314	~	7
276	00001202766 01	C	ΙX	1,2,	303	00000303	9	238
276	63400203573 01	0	Š	201,	303	00000303	~~ 1	Ŋ
276	63400203617 01		Q	٥2	303	00000303	10	240
277	300000202703 01	329	×	7	303	00000361		7
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277	10 900000000100	231	SX	MPRINIOH	303	10000304	_	7
277	007400015451 01	e	×s	CH	303	00000361	10 240	7
277	007400003614 01	c	×s	I×	303	00000314		7
02774	+007400003615 010	0	×	I X2	304	10000361	~~ ↓	7
277	007400003147 01	0	SX	2)+5	304	00000361	-	
277	007400003602 01	0	SX	N2	304	50000004		m
277	07400403110 01	C	×s	02,	304	10000315		7
300	053400203570 01	C	O _X	C)102,2	304	00000000		CLM
300	53400403614 01	C	ΩX	IΧ°μ	304	3000000	0	LLS 18
300	050000003620 01	0 232A	۲V	I PUNCH	304	30000355		

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AERODYNAMIC	03047 +050000003614 010 CLA IX 03050 +040000003551 010 ADD 1)+1	051 +013100000000 00 XC	052 +020000003142 010 MPY	053 +076700000021 00 AL	3054 +040000003142 010 ADD 2	3055 +040000003613 010 ADD J	3056 +060100003612 010 STO J	3057 -050000003602 010 242A CA	3060 -063400403170 010 SXD 6)	3061 +007400400004 010 TSX (ST	3062 +000000003257 010 PZE 8	3063 -053400403170 010 LXD	3064 +056000003613 010 243A LD	3065 -1000000000000 00 STR	3066 +056000003612 010 LD	3067 -10000000000000 00 ST	3070 -063400403170 010 SXD 6	3071 +007400400005 010 TSX	3072 -053400403170 010 LXD 6)	073 +002000003100 010 TRA 244	3074 -053400103563 010 D)32E LXD C)G	3075 -053400203570 010 D)22E LXD C	3076 +002000003100 010 TRA 244	3077 -063400403566 010 E)12E SXD C)G	244A BS	100 +100001103101 010 244A1 TXI *+	101 -063400103563 010 SXD C	102 -300000100407 010 244A2 TXL 96	103 -0500000003601 010 245A CAL	104 -063400403170 010 SXD 6)	105 +007400400010 010 TSX (106 -053400403170 010 LXD 6)	107 +0020000000000 010 246A TRA 2	110 +0500000003167 010 A)102 CLA 6	060100003553 010 STO 1!+	112 +050000003614 010 CLA IX	113 +040200003167 010 SUB 6

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03162	-206060233031	000		100	-206060233031	03230	-206001063060	00	BCD 1 16H	
3164	233000000000	0	× 39	\circ	3300000000	3232	206060606060) ر	
3165	77777700000	0		ں	777500000	3233	2060606060	000	۵ ۵	
3166	000000000000	00		Ç	00000000000	3234	222551604626	00	O 1BER	
3167	000001000000	00		U	000010000	3235	03060456444	00	D 10H N	
3170	000000000000	00		ပ	0000000000	3236	10406677301	00	0 1/46	
3171	000000000000	00		ပ	00000000000	3237	250105331061	00	D 1E15.8	
3172	213460606060	00		Ç	**	3240	36142511301	00	D 1./KR=	
3173	052151706061	00		ပ	NARY	3241	77307306001	00	D 1X,7H	
3174	303144212731	00		S	IHIMAGI	3242	340130600411	00 8)2L	D 1(1H	
3175	200101677311	00		Ç	11X,	3243	010501310334	00	D 115113	
3116	043051252143	00		C	4HREA	3244	05164603044	00	D 1HRU H	
3177	306001006773	00		C	Y 10X	3245	00100306063	00	D 1 10H	
3200	212731452151	00		C	AGINA	3246	09090909090	00	۵	
3201	206011303144	00		S	1H6	3247	09090909090	00	٥	
3202	206060606060	00		C		3250	09090909090	00	0	
3203	206060606060	00		C		3251	336060606060	00	BCD 1,	
3204	010067736060	00		C	10X	3252	40105013103	00	O M	
3205	305125214360	00		ပ	1HREAL	3253	54662336030	00	D INOS.	
3206	200101677304	00		C	1X o	3254	32151246260	00	D 1CARD	
3207	273145215170	00		C	INAR	3255	52330252460	00	D 1NCH	
3210	331130314421	00		C	WIH6	3256	020430604764	00	D 124H	
3211	214360010167	00		C	111	3257	46103106773	00 8175	D 1(/38X	
3212	277304305125	00		S	X, 4	3260	22561613460	00	D 1SE//}	
3213	340130600102	9 00	NS C	C	1H 1	3261	212470602321	00	D 1ADY C	
3214	216225616134	00		C	E//	3262	32560626325	00	D THE ST	
3215	206005306023	00		C	π	3263	30106306063	00	D 1,16H	
3216	206060606060	00		C		3264	036161041167	00	D 13//4	
3217	206060606060	00		C		3265	36260130131	00	0 1TS =1	
3220	306060606060	00		ů		3266	36047463145	00	D 1L POI	
3221	034321634651	00		C	LATO	3267	34645635146	00	D 1 CONTR	
3222	256046622331	00		S	OS	3270	06001073060	00	0 1 17	
3223	010630606330	00		S	± H9	3271	06060606060	00	BCD 1	
3224	216104106773	00		C	748X	3272	06060616360	00	BCD 1	
3225	236213013103	00		C	115=113	32	060656060	00	BCD 1	
3226	36047463145	00		C	POI	3274	2551604626	00	0	
3227	234645635146	00		8C0	72	32	03060456444	00	BCD 10H NUM	

762 PAGE 34	D 10H NU	D 1/46X9	D 1E15.8	D 1./KR	D 1X o 7H	4 HI)1 O	0 1	D 13X,1	D lill49	0 1113	۵	D IIIE12	D 18}}	D lolE15	D 1.81	D 1 1E1	Δ	۵	O	D 12Xp	D leli4,	D 11/1(23	BCD 1XN(1),	INIOHI O	D INTROL	D 120H	D 1 14X	_	; ∩	DIXCI	D 1 DELT	D 17X,10	D I NO.	D 10F PT	D I CONT	BCD 1X8	IH I) I O
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INFLUENCE COEFFICIENTS	3344 +00306045644	3345 -21040667730	3346 +25010533106	3347 +33614251130	3350 -27730730600	3351 -34013060041	3352 +34606060606	3353 +03677301310	3354 -34013104730	3355 +01310334606	3356 +10730605677	3357 -34012501023	3360 +10343460606	3361 -33012501053	3362 +33107301036	3363 -20600125010	3364 -20606060606	3365 -20606060606	3366 -20606060606	3367 +02677360606	3370 -33013104730	3371 -21617402036	3372 -27457431347	3373 -07463145636	3374 -05635146436	3375 +02003060234	3376 -20600104677	3377 20606060606	3400 -20606060606	3401 -27743134606	3402 20242543632	3403 +07677301003	3404 -20454633600	3405 -06436047633	3406 -20234645635	3407 -27736001063		3411 ~20616134606
AERODYNAMIC	CD 1/46X,	CD 1E15.8	CD 1°/KR≖	CD 1X,7H	CD 1(1H 4	CD 11X	CD 11 MAT	CD 1THE C	CD 1X,15	CD 1(//	0	CD IRIX /	CD 1HD MA	CD 1 1	CD 19X,15	BCD 1(//	CD 1IX /	CD IS MAT	CD 1THE C	CD 1X,15H	BCD 1(1H	CD IASE !!	CD 1 5H	0	2	CD 1	CD INS	CD THE TR	CD 1,14H	CD 13//49	CD TS = 1	CD IL POI	CD 1.CONTR	CD 1 17	3	ပ		ပ
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	3276 -210406677301 0	3277 +250105331061 0	3300 +336142511301 0	3301 -277307306001 0	3302 -340130600411 0	3303 +316760616134 0	3304 +316044216351 0	3305 -233025602330 0	3306 -277301053060 0	3307 -346161600411 0	3310 +34606060600	3311 -113167606160 0	3312 +302460442163 0	3313 ~206330256023 0	3314 +116773010530 0	3315 -346061616004 0	3316 +316760616034 0	3317 -226044216351 0	3320 -233025602330 0	3321 ~277301053060 0	3322 ~340130600411 0	3323 +216225616134 0	3324 -206005306023 0	3325 -206060606060 0	3326 206060606060 0	3327 ~206060606060 0	3330 -056231254563 0	3331 +302560635121 0	3332 -330104306063 0	3333 +036161041167 0	3334 -236260130131 0	3335 ~036047463145 0	3336 +234645635146 0	3337 -206001073060 0	3340 -206060606060 0	3341 -206060606060 0	03342 -206060606060 00	3343 +222551604626 0

	AERODYNAMIC	INFLUENCE COEFFICIENTS 10/11/62	/62 PAGE 35
12 -236024216321 00 13 -204664634764 00	BCD 1T DATA	03460 +013104730111 00	BCD 1114,19 BCD 1 47X
4 +026773010230 0	CD 12X,12	462 -2060606060	·
5 -346061616005 0	CD 1(//	463 -206060606060	BCD 1
6 +331034346060 0	CD 1.8)	3464 -226160606060	5
7 -277301250105 0	CD 1X, 1E1	3465 +252363314645	1ECT10
0 +053310730110 0	CD 15.8	3466 +047311306062	
1 +056773012501 0	CD 15X,1E	3467 +040767730131	147Xº1
2 -206060607403 0	CD 1 (3470 +242163216161	1DATA/
3 -206060606060 0		3471 +314547646360	N I
4 -206060606060 0		2 -277301013060	1X, 11
5 -206060606060 0		3473 -340130600411	1 (]
-216060606060	CD 1/	3474 -216134606060	BCD 1//)
7 -202151252161 0	CD 1 A	3475 -233025465170	1THE
0 -204346232143 0	CD 1 CO	3476 -002246247060	1-80D
1 -277301023060 0	CD 1X,	3477 -032545242551	LEN
2 -343134730204 0	CD 1(I),2	3500 +265146446062	1 FROM
3 -27730.5306067 0	CD X 5H	3501 -206002053060	7
4 -340130600311 0	CD 1(1H 3	3502 -206060606060	BCD 1
5 -216134606060 0	CD 1//3	3503 -206060606060	7
6 +012501053310 0	CD 11E15.	3504 #312545636260	1 I EN
7 -206251252613 0	CD 3 SREF	3505 -062526263123	10EF
0 +041167730630 0	CD 149X96	3506 +254523256023	IENC
1 +250105331061 0	CD 1E15.	3507 -203145264364	Z H
2 +033060621301 0	CD 13H S=	3510 -304521443123	YNA
3 +106105026773 0	CD 18/52X	3511 -202125514624	
4 +130125010533 0	CD 1=1E1	3512 +116773030530 00	8CD 19X,
5 -330430602251 0	CD 1,4H B	3513 -340130016002	1 (1H
6 -206061050167 0	CD 1 /51	3514 -216134606060	11
7 -206060606060	00	3515 -233025465170	1 THE
0 -206060606060 0		3516 -002246247060	1-8
1 +010533106060 0	CD 115	3517 -032545242551	1 LENDE
2 +222151130125 0	CD 1BAR=1	3520 +265146446062	hom
3 -277306306023 0	CD 1X,6H	3521 -206002053060	CD 1 2
4 +256261610411 0	CD 1ES//	3522 -206060606060	S
5 -034623316331 0	CD 1LOCIT	523 ~206060606060	
6 +232524606525 0	CD 1CED V	524 +254563626060	CD JENT
7 +306051252464 0	CD 1H RED	525 +252626312331	CD 1E

MPRINT

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MPRINT03
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                                                                                                                                                                                                                                                                                                                                                                                                                     MPRINT39
                                                                                                                                                                                                                                                                                                                              MPR INT31
                                MA = DIMENSIONED NUMBER OF ROWS
NTAPE = TAPE NUMBER FOR PRINTING
                                                                                                                                                                                                                                                                                                                                                                          (C(K) , K=1, N2)
                                                                                                                                                                                                                                                                                  (IT(I), I=1,N2)
                                                                                                                         114
(A,M,N,MA,NTAPE)
                                                                 SUBROUTINE MPRINT (A,M,N,MD,NTAPE )
                                                                                                                       (1H 5 4X, 6( 6X, 7HCOLUMN (1H 114, X) (6E 17.8)
                   MATRIX TO BE PRINTED NUMBER OF ROWS NUMBER OF COLUMNS
                                                                                                                                                                                                                                                                                                                                                                        WRITEOUTPUTTAPE NTAPE, 3, 1,
                                                                                       A(1), IT(6), C(6)
                                                                                                                                                                                                                                                                                 WRITEOUTPUTTAPE NTAPE, 2,
SUBROUTINE MPRING
                                                                                                 (IT,C)
                                                                                                                                                                                                                                                                                                                                                                                  IF (N3-N1) 10,11,11
                                                                                                                                                                                                     6,6,5
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                                                                                                                                                                                                                                                  I= N4,N3
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                                                                                                                                                                                                                                                                                                                                                  C(K)=A(L)
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                                                                                                 EQUIVALENCE
                       11 11
                                                                                                                                                                                                    IF (N3-N1)
                                                                                                                                                                                                                                                                                                                                                              L=L+MD
                                                                                                                                                                                                               N2=N1-N3+6
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MPRINT40

MPRINT

11 RETURN END(1,1,0,0,0,0,0,1,0,1,0,0,0,0)

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PAGE	
1,62	
10/11	

MPRINT

STORAGE NOT USED BY PROGRAM

	DIMENSION AND EQUIVALENCE STATEMENTS	DEC OCT	DIMENSION, OR EQUIVALENCE STATEMENTS	DEC 0CT N1 182 00266	AM FORMAT STATEMENTS	EFN LOC	NG IN SOURCE PROGRAM	DEC 0CT 9) 173 00255 E)E 139 00213	ER VECTOR	DEC OCT	OUTPUT FROM LIBRARY
	DIMEN	100	DIMEN	00267 00267 00263	PROGRAM	707	APPEARING	00232 00232 00262	TRANSFER	OCT)T 0UT
	LES APPEARING IN	DEC	APPEARING IN COMMON®	DEC L 183 N4 179	ITIONS FOR SOURCE	A P	OTHER SYMBOLS NOT AP	DEC 6) 154 C)202 178	OF NAMES IN	DEC	TO SUBROUTINES NOT
0CT	FOR VARIABLES	0CT 00277	NOT APPE	00270 00270 00264	OLS AND LOCATIONS	LOC 00244	FOR OTH	00227 00227 00261	LOCATIONS	00000	POINTS TO
DEC OCT 32561 77461	S	DEC 191 0	BLES NO	DEC 184 C 180 O	BOLS AN	EFN 3 0	SNOI	DEC 151 0 177 0		DEC 0 0	ENTRY P
· · ·	STORAGE LOCATION	I	NS FOR VARIABLES	X W	SYMB	т 80	LOCAT	2) C) 62		(STH)	-
00£00	STOF	00277	STORAGE LOCATIONS	00271 00271 00265		LOC 00254		00256 00256 00260		0000 1	
DEC 192		DEC 191	RAGE L	DEC 185 C 181 C		EFN 2		DEC 174 (176 (DEC 0CT 1 00001	
		u	STO	N N		8 3 2		C) C)		(FIL)	

(STH)

NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS
AND
NUMBERS
FORMULA
INTERNAL
CORRESPONDING
WITH
NUMBERS
FORMULA
EXTERNAL

LOCATION	
AND OCTAL	EFN IFN LOC 7 17 00071 11 40 00223
NUMBERS	EFN I 7 11
FORMULA	4 3
INTERNAL	EFN IFN LOC 6 14 00053 10 37 00214
UMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATION	EFN I 6 10
H	N LOC 12 00045 30 00161
NUMBERS	1FN LOC 12 00049 30 0016
FUKMULA	EFN 5
EXIEKNAL FURMULA	N LOC 11 00040 29 00153
	EFN IF

000046 000047 000050 000051 000052 000054	+0402000022 +04000000022 +06010000026 +05000000026 +06010000023 +06010000023	010 010 010 010 010	13A 14A	SUB ADD STO CLA STO CLA STO	NON NON XX
000055 000056 000057 000060 000062 000065	-05340010027 -06340010026 -05340020026 +05000000026 +06220000007 -075400200007 +06010000027 +06010000023	010 010 010 0010 010 010	15A 16A	LXD SXD CLA STD STD STD STD STD	**************************************
00070 00071 00072 00073 00074	-06340040026 +05000000027 +06010040030 +10000120007 -06340020027 -3000020000	010 011 010 010	17A 17A1 17A2	SXD CLA STO TXI SXD	C 5 6 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7
00016 000100 00100 00101 00103 00105 00106 00110 00111	-0500000000000000000000000000000000000	000 0010 0010 0010 0010 0010 0010		S X X X X X X X X X X X X X X X X X X X	NIAP (S)+4 (STH (S)+1 (N) (N) (S)+1 (N) (S)+1 (S)+1 (F) (F)(F) (F)(F)

1 Z вср PZE 97d SXD SXD CLA ADD STA CLA STA CLA CLA STA STA CLA STA STA CLA STO CLA STO BCD CLA STO CLA STO CLA STO SUB TZE TPL (STH) (FIL) 1141 10A 11 12A Y T ZA 84 V6 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 010 00 00 00 00 00 00 00 -346263303460 -342631433460 +000000000000+ +062100000030 000000000000+ 000000000000+ -044751314563 -063400100002 -063400200003 -063400400004 +040000000255 +050000400003 +062100000154 +062100000076 +060100000266 +060100000265 +05000400001 +062100000151 +050000400002 +062100000117 +05000400004 +062100000132 +05000400005 +050000000000 +060100000237 +0500000000000 +050000000227 +060100000264 +050000000230 +060100000263 +050000000264 +040200000266 +010000000053 +01200000045 +05000000256 +062100000161 +050000000227 +00200000053 MPRINT 00000 00002 50000 **40000** 90000 00000 00012 00013 00014 00016 00020 00025 00026 00034 20000 70000 00011 00015 00022 00023 00024 00000 00031 00032 00033 30035 96000 00040 10000 00043 0000 00017 00021 00042 20044 90000 00027 00037

PAGE 6	SXD 6)+4,4 TSX (STH),4 PZE 8)3 LXD 6)+4,4 LDQ 1 STR LXD 2)+1,1 CLA N2 STD 33A2 LDQ C+1,1 STR TXI *+1,1,1 TXL 33A,1 SXD 6)+4,4 LXD C)202,1 TXI *+1,1,1 SXD C)202,1 TXI *+1,1,1 SXD C)202,1 TXI *+1,1,1 SXD C)202,1 TXL 24A,1 CLA N3 SXD C)62,4 CLA N3 SUB N1 TXL 24A,1 CLA N3 SUB N1 TXL 24A,1 CLA N3 SYD C)62,4 CLA N3 SYD C)62,4 CLA N3 SYD C)62,4 CLA N3 SYD C)62,4 CLA N3 SYD C)62,4 CLA N3 SYD C)62,4 CLA N4 ADD 2) STO N3 CLA N4 ADD 2) STO N4 TXL 11A LXD \$+1,2 LXD \$+1,2 LXD \$+1,2 LXD \$+1,2 LXD \$+2,4 TRA 6,4
10/11/62	00162 -063400400236 010 00164 +00000000244 010 00165 -053400400236 010 00165 -053400400236 010 00171 -053400100230 010 32A 00171 +050000000000 01 00172 +06220000176 010 32A 00173 +056001100176 010 33A1 00174 -100000000000 01 00177 -063400400236 010 35A1 00175 +100001100173 010 33A2 00177 -063400400236 010 00201 -053400100262 010 00202 -053400100262 010 00203 +100001100264 010 35A1 00205 -063400100262 010 00206 -300000100123 010 35A1 00207 +05000000264 010 36A 00211 +010000000264 010 37A 00212 +012000000264 010 37A 00213 -063400400261 010 E)E 00214 +050000000264 010 37A 00215 +040000000263 010 00217 +050000000263 010 00217 +050000000263 010 00217 +050000000263 010 00217 +050000000263 010 00217 +050000000263 010 00217 +050000000263 010 00217 +050000000263 010 00221 +00000000263 010 00221 +000000000263 010 00221 +000000000000000000000000000000000
	LXD 6)+4,4 LXD 2)+1,1 SXD C)202,1 CLA M STD 35A2 PXD 0,1 STO 1 CLA 2)+2 STO K,2 SYD C)60,2 CLA N4 SYD C)60,2 CLA N4 SYD C)62,4 LXD MD MPY 1)+1 ADD 1 STO L LXD N4,2 CLA N3,2 CLA N3 SYD C)62,4 LXD N4,2 CLA N3 SYD C)62,4 LXD N4,2 CLA N3 SYD C)62,4 LXD C)62,4 LXD C)62,4 LXD C)62,4 LXD C)62,4 LXD C)62,4 LXD C)62,4 LXD C)62,4 LXD C)62,1 CLA K ADD 2)+1 SYD C)60,1 CLA C LXD C+1,1 SYD C+1,1 CLA L SYD C+1,1 CLA L SYD C+1,1 SYD C+1,1 SYD C+1,1 SYD C+1,1 CLA L SYD C+1,1 SYD C+1,1 CLA L SYD C+1,1 CLA L SYD C+1,1 CLA L SYD C+1,1 CLA L SYD C+1,1 SYD C+1,1 CLA L SYD C-1,1 SYD C-1,1 SYD C-1,1 SYD C-1,1 SYD C-1,1 SYD C-1,1 SYD C-1,1 SYD C-1,1 SYD C-1,1 SYD C-1,1
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1 LUMN

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PAGE	*		
10/11/62	######################################		LOC OF ARRAY WGRD CGUNT END=0 IF TRANSFER CARD
S ON TAPE. FIBII	**************************** SEQUENCE X		X1,1 X2,2 6,4 14D 1,4 AYRAY ARRAY ENO
E COL BIN CARDS	**************************************	(10S) (WRS) (RCH) (MTC) (WER)	BINPU SXA SXA CAL* STD CLA STA CAL*
BINPU ROUTINE TO WRITE	90000	TRANSFER VECTOR 00000 743146623460 00001 746651623460 00002 745123303460 00003 746663233460 00004 746625513460	00006 0634 00 1 00142 00007 0634 00 2 00143 00010 -0500 60 4 00006 00011 0622 00 0 00331 00012 0500 00 4 00001 00013 0621 00 0 00062 00014 -0500 60 4 00002

ie 2	BINPU030 BINPU031 BINPU032 BINPU033 BINPU035		1	BINPU049 BINPU050 BINPU051 BINPU053 BINPU053	8 INPU055 8 INPU056 8 INPU057 8 INPU059 8 INPU060 8 INPU061	BINPU063 BINPU064 BINPU065 BINPU066
PAGE		* *				
10/11/62	CONTROL WORD	ADD RELATIVE BIT 7-9, WORD COUNT=22 CONTROL WORD ESTABLISHED. ************************************	TO ITS NORMAL STATE 4TH, 5TH ARGS	TSXES IS IS ID FLAG BLANK SEQ. NO. NO NON-ZERO.	S IS SEQ NO. Q NO TO BCD	NEXT ARGUMENT 2 EXTRA ARGS. COUNT ZERO A TRANSFER CARD
	UP CON	ADD RELATIVE 7-9, WORD COUCONTROL WORD ************************************	BLSEQ FOR 4		THIS	T0 38T 3RD 8E
	SET (ADD RELAT 7-9, WORD CONTROL W ************************************	SET BITEST	NO MORE BIG, TH EQUAL, IS SEQ	NO SMALL, CGNVERT SAVE	MOVE TO AT MOST IS WORD MUST BE
FIBII		REL IMAGE 7-9, WORD C CIMAGE CONTROL WOR ************************************				
TAPE.	D1 LCCN CCUNT,0 3,4 18	REL IMAGE CIMAGE ****** FOURTH E WHETHE	2.5 BLSEQ 4,4 MSKPDT MSKTSX	62 4,4 618 63 BLSEQ *+2	*+2,4 COSEQ,4 **,4 SEQNO GS,4,-1 BCDID	65,4,-1 64,2,1 X4,4 END TRCD
ARDS ON	100 000 3,44 18	REL INA INA INA CINA CINA CINA ERMINE SET CEL	2,2 2,2 BLSE 4,4 MSKP	62 4,4 6,16 63 63 8LSE *+2	* 0 * 0 0 0	200×m+
N I	SUB STD SXA CLA*	0RA 0RA 0RA SEW + + EST DETER AND S	AXT STL CAL ANA ERA	TNZ CLA* LAS TRA STZ TNZ	P X X X X X X X X X X X X X X X X X X X	TXI TIX SXA NZT TRA
COL 8		* * * *			<u>ش</u>	625
O WRITE	00325 00066 000061 00003 000022	026 026 033 174	000 030 000 026 030	0005 0000 026 0005 0000	000000 00000 000172 00000 00053 00053	0005 0003 014 777 015
ĕ 10	000000				000000000	
ROUTINE	00040	0000	4 10 0 0 0	000000	4 0 4 4 7 6 6 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	777 000 4 0 0 0
	040 062 063 050 077	000	071 -062 -050 -032 032	000000	-075 -013 -013 -063 -063 -1060 -1060	1 1 2 C C C C C C C C C C C C C C C C C
BINPU	00016 00017 00020 00021 00022	000	~ N M M M M	~ t t m m m m	00043 00044 00045 00046 00047 00050	, ro, ro, ro, ro, ro,

PAGE 3

10/11/62

	* * * * * * * * * * * * * * * * * * * *	¥ 7 ¥	**************************************	*************************************	BINPU068 ***BINPU069 BINPU070 ***BINPU071
0057 0774 00 2 0002 0060 -0754 00 0 0000	NEXT	AXT PXD	22,2	CLEAR AC FOR CHECKSUM.	BINPUO72 BINPUO74 BINPUO74
00061 0774 00 4 00000 00062 0560 00 4 00000	COUNT	LDO	0 20 U U V W H U	MOVE ARRAY INTO CORE.	BINPU075 BINPU076
0064 0361 00 2 7777 0065 1 00001 4 0005		ACL TX1	CIMAGE+24,2 CIMAGE+24,2 *+1,4,1	ACCUMULATE CHECKSUM FOR BODY.	BINPUOT8
0066 3 00000 4 0016 0067 2 00001 2 0006	LOCN	TXH	OUT,4,** ARRAY,2,1	FINISH WHEN SPECIFIED BY NO. WORDS DESIRED.(2,4)	BINPU080 BINPU081
0070 0634 00 4 0006	Z.	SXA	COUNT # #	SET COUNT FOR NEXT LOOP.	BINPU082
0072 0602 00 0 7774		SLW	CIMAGE+1		BINPU084
	****	****	*************	***************************************	BINPU085
	*	EDIT	EDIT THE IDENTIFICATION FIELD	N FIELD.	BINPUOST
	* * *	****	**************	经运货条件 化苯基苯基苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯苯基苯基苯基苯基苯基	***BINPU088
0073 -0500 00 0 0026	EDIT	CAL	SEONO		BINPU090
0074 0560 00 0 0032		100	L(1)		BINPU091
0016 - 0500 00 0 0030		CAL	BCDID		BINPU093
0000 0 00 0 0000		197	9	•	BINPUOST
0100 -0600 00 0 0032 0101 -0130 00 0 0000		o io XCI	IDECO	SAVE FOR FINISHING.	BINPU095 RINPHO96
0102 0634 00 1 0012		SXA	SVI,1		BINPU097
0103 0774 00 2 0000		AXT	4,2		BINPU098
0104 0774 00 4 0000 0105 0774 00 1 0000	ABC	AXT AXT	2,4 3,1		BINPU099
0106 -0754 00 0 0000		P XD			BINPULOI
0107 -0114 01 0 0023 0110 -2 00001 1 0011		CAD	TAB, 1 *+3,1,1		BINPU102 BINPU103
00111 0767 00 0 00014 00112 0020 00 0 00107		ALS TRA	12		BINPU104 BINPU105

PAGE 4	BINPU106 RINPU107	BINPUIOS	BINDITIO	BINPUIL	1	**************************************	BINPU115	OTTOJNI OZNAZAZA	BINDOILS	BINPU119	BINPU120	BINPUIZI	BINPU122	BINPUIZS BINDUIZS	BINPO124	BINPUIS	BINPU127	BINPU128	BINPU129	BINPU130	BINPUIST	BINPUI32	BINPOISS	BINPOIS4	CCIOLNIZO	BINPULSO	BINPUISB	BINPU139	*******BINPU140	BINPU141	BINPU142	BINPU143
I 10/11/62	COL BIN AT LAST TO LAST+3	FINISH W/SAVED C(MO).			化光环 计光光 医电电电电电电电电电电电电电电电电电电电电电电电电电电电电电电电电电电	IS BUILT, WITH THE BODY	+23, AND ID AT LAST THRU LAST+3. CARD ON TAPE. *******	*********************************			ESTABLISH 1/0 FOR TAPE 14.			SET (WER) FOR RETRY.				INCREMENT CARD COUNT.		TECT IE IACT CADO	NOT THE LANT CARD			ALL DONE, EXIT					在本本市市市中市市 1991年1991年1991年1991年1991年1991年1991年19	UPDATE THE CARD URIGIN.		
DS ON TAPE. FIBII	LAST+4,2 *+1,2,-1	ABC,4,1 IDLCD	ABC-1,2,0	***	************	ENTIRE CARD IMAGE IS	MAGE THRU CIMAGE+23, ***** WRITE THE CARD	**********		140	(TOS)	PUNCMD.4	\$ (RCH)	↑•0	_	5 (WER) , 4		L111	SEDNO) U	SWTCH	BPTES	\$ (TES)	**,1	**,2	5.**	2,4	2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	**************************************	A22	HUVALO	;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;;
BIN CARDS	SLW	11X	HXL	AXT	*	111		*	NOP.	CAL	× X	AXC	XEC*	PXA	STA*	× 2.	רא ס	200	N IS	ZET	TRA	CAL	*MTS	AXT	AXT	AXT	A X	****		ACL	MIS	:
COL			i	2 N T	* * * *	•	• •	* * * *							į	BPIES								7	X2	×		***	WTC)		
BINPU ROUTINE TO WRITE	00113 0602 00 2 77734 00114 1 77777 2 00115	0116 0560 00 0 0032	0117 3 00000 2 0010	0150 0114 00 1 0000					00121 0761 00 0 00000	0123 0074 00 4 0000	0124 0522 60 0 0000	0125 -0774 00 4 0021	0126 0522 60 0 0000	3127 0754 00 4 000C	0130 0621 60 0 0000	132 -0500 00 00 0000	0133 0400 00 0 0030	0134 0114 06 0 0021	0135 0602 00 0 0026	7777 0 00 0250 010	1137 0020 00 0 0014	0140 -0500 00 0 0013	0141 0602 60 0 0000	142 0774 00 1 0000	145 0114 00 2 0000	144 07/4 00 4 0000	0000 + 00 0300 0+		0146 -0500 00 0 7774	00147 0361 00 0 00333	0150 0602 00 0 7774	

GE 5	BINPU144	BINPU146	BINPU147	BINPU148	BINPU149	BINPU150	BINPUI51	BINPU152	BINPU153	***BINPU154	81NPU155	BINPU156	BINPU157	BINPU158	BINPU159	BINPU160	BINPU161	BINPU162	BINPU163	BINPU164	BINPU165	BINPU166	***BINPU167		BINPUL99	BINPUITI	BINPU172	BINPU173	BINPU174	BINPU175	BINPU176	2 TOANTO	BINPOI/8	BINPU180	BINPU181
PAGE	* * * *									*													****	,	×										
10/11/62	A NEXT 市本年本市市市市市市市市市市市市市市市市市市市市市市市市市市市市市市市市市市		CLEAR REST OF	CARD IMAGE.		STORE ZERO WORD COUNT				***************************		WORD COUNT EXHAUSTED	RETURN IF CARD IS FULL	NO.N	CORRECT WORD COUNT			RETURN CHECKSUM.		CLEAR REST OF CARD.			***********	A BINARY INTEGER TO BCD. (4 DIGITS DECR-MQ)	李龙帝李本女女心看着女女女女女女女女女女女女女女女女女女女女女女女女女女女女女女女女女女		TEST IF BLANKS DESIRED.		RIGHT ADJUST BIN INTEGER						
DS ON TAPE. FIBII	NEXT		23,2	CIMAGE+24,2	+-1,2,1	ZWC	CIMAGE	EDIT		*****		END	INIC	COMMON	0,2	CIMAGE	CIMAGE	COMMON	IN,2,0	CIMAGE+24,2	*-2,2,-1		*******	OUTINE CONVERTS A BINAR	***************************************			COSEQX	18	TEN	COMMON	i L	- v	COMMON	
COL BIN CARDS	TRA *******	: *	TRCD AX	-	-	_	STD	α.	*	******	*	OUT STZ	XNT	SLW	PXD	SUB	STD	CAL	TXL	STZ	IXI		*	THIS RO	**	~	2N	\propto	œ	~	- >	Κ.	0 V	ıα	~
BINPU ROUTINE TO WRITE	0151 0020 00 0 00057		152 0774 00 2 0002	153 0600 00 2 7777	154 2 00001 2 0015	155 0500 00 0 0032	0156 0622 00 0 77740	157 0020 00 0 0007				160 0600 00 0	161 -2 00001 2	162 0602 00 0	163 -0754 00 2	164 0402 00 0	165 0622 00 0	77777 0 00 00 00 00 00 00	167 -3 00000 2	170 0600 00 2	171 1 77777 2					172 -0754 00 0 0000	173 -0520 00 0 0030	174 0020 00 0 0021	175 0765 00 0 0002	176 0221 00 0 0033	7777 0 00 1 00 0 7777	200 -0754 00 0 0000	0201 0221 00 0 00332 0202 0767 00 0 00006	203 -0602 00 0 7777	204 -0754 00 0 0000
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PAGE 6	BINPU182 BINPU183 BINPU184 BINPU185 BINPU186 BINPU187	*****BINPU189 BINPU190 BINPU191 BINPU192 BINPU193 BINPU193	BINPU195 BINPU194 BINPU198 BINPU199 BINPU202 BINPU203 BINPU204 BINPU205 BINPU205 BINPU205 BINPU206 BINPU207 BINPU207 BINPU208 BINPU209	BINPUZIO
10/11/62	•	· · · · · · · · · · · · · ·		
	RIGHT ADJUSTED SEQ NO. BLANK OUT SEQ. NO.	.*************************************	0 2 3 4 5 6 6 7 7 8 9 0 WITH CARRY CONVERSION CONSTANTS ,100,40,20,10,4,2,1	,42
CARDS ON TAPE. FIBII	TEN 12 COMMON 1,4 BLANK 1,4	CIMAGE,0,24 LAST,0,3 CD ADDITION OF 1	TB1 TB1 TB1 TB1 TB1 TB1 TB1,0,4096 TB1 TB1,0,4096 TB1 TB1,0,4096	77777770000,102,42
COL BIN	DV AL OR TR OSEQX CA	PUNCMD 10CP 10CD 10CD 10CD 10CD 10CD 10CD 10CD 10CD	* TB1 HTR TB MPY FAD ADD CLA STZ CPY TXI TXI HTR * TABLES FOR TAB TAB TAB TAB TAB TAB TAB TAB TAB TAB	MSK2CH OCT
OU ROUTINE TO WRITE	0221 00 0 00332 0767 00 0 00014 -0501 00 0 77777 0020 00 4 00001 -0500 00 0 00306 0020 00 4 00001	-0 00030 0 77740 0 00003 0 77730	0000 00 0 00215 0100 00 0 00215 0200 00 0 00215 0300 00 0 00215 0400 00 0 00215 0500 00 0 00215 0500 00 0 00215 1 00000 0 00215 1 10000 0 00215 1 10000 0 00215 1 10000 0 00215 1 00000000000000000000000000000000000	9
BINPU	00205 00206 00207 00210 00211	00213 00214	00215 00216 00217 00220 00221 00223 00224 00224 00225 00226 00231 00233 00233 00233 00233	0242

PAGE 7		BINPU211		BINPU212								P I CHON I S	B INDICATA	170 11170	BINPU215	BINPU216	BINPU217	BINPU218									BINPU219	BINPU220		BINPU221	BINPU222	BINPU223	BINPUZZ4
10/11/62 PA				1,4010,4004,4002,4001														,2010,2004,2002,2001															10,1004,1002,1001
ON TAPE. FIBII		0,0,0		4000,4400,4200,4100,4040,4020,4010,4004,4002,4001								8.1	4102.4042	1	0,7,-1		1,	2000,2400,2200,2100,2040,2020,2010,2004,2002,2001										2102,2042		1,	- (0 1 000 1 300 1 300 1000 1000 100	0,1400,1200,1100,1040,1020,1010,1004,1002,1001
E COL BIN CARDS		ID123 OCT		100									4		-	⋖	SEONO BCI	\Box									BLSEQ	T)()		BCDID BCI	BLANK		ר ז
BINPU ROUTINE TO WRITE	00243 +000000000102 00244 +0000000000042	10 0	~	\sim	00252 +000000004400	1 ~~	.+	ia	∽ ►	_ ~	` -	. ^	1 N	٠.+	5 -3 77777 7 0000	•	~	\sim	_	ΛI	∾ .		.∧	~ ∸	\sim	+000000005001	\sim 1	∾ .	. +	10	5 606060606060	_ ^	$\overline{}$

PAGE 8		BINPU225	8INPU226	BINPU227	BINPU228	BINPU229	BINPU230	BINPU231	BINPU232	BINPU233	BINPU234	BINPU235	BINPU236	BINPU237	BINPU238	BINPU239
10/11/62		ZERO WORDS FOR TCD									CONTROL WORD SKELETON					
CARDS ON TAPE. FIBII		00020000200	1102,1042	0,0,1		1	5	16,,12	10	22	000526000000	-1	-32	-40	COMMON-1	
BIN CA		100	OCT		PZE	PZE	P2E	PZE	DEC	I	OCT	Ш	ш	ш	SYN	END
COL		ZMC		DJ	IDLCD	L(1)	5 A	140	1 EN	A22	IMAGE	COMMON	CIMAGE	LAST	END	
BINPU ROUTINE TO WRITE	00311 +000000001400 00312 +000000001200 00313 +000000001100 00314 +000000001040 00315 +000000001020 00317 +000000001010 00320 +000000001002	00322 +000500000000	00323 +000000001102 00324 +000000001042	00325 0 00001 0 00000	00326 0 00000 0 00000	00327 0 00000 0 00001	0 00000 0 00000		+00000000012	0000 00 0 00026	00334 +000526000000	77777	77740	77730	77776	

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII POST PROCESSOR ASSEMBLY DATA

335 IS THE FIRST LOCATION NOT USED BY THIS PROGRAM

	225,
	224,
	223,
	222,
	221,
335	220,
160,	217,
136,	216, 205 335
80LS 52 . 167 122 117 55\$	2156 2014 214 ₈ 133 67 76
DEFINED SYMBOLS 5A D1 16 G2 34 G3 37 G4 53 G5 50, 52 IN 161, 167 X2 7 X2 7 X4 54 14D 11, 122 A22 147 A8C 115, 117 END 15, 55	102 107 134, 176, 155 36 157 113, 17, 17, 161, 151, 56, 13,
TO DEFIN 5A 01 62 63 64 65 1N X1 X2 X4 14D A22 ABC END	SV1 TAB TB1 TEN ZWC BCI8 EDIT LAST LOCN L(1) NEXT TRCD ARRAY BCDID BINPU
REFERENCES 330 325 54 51 51 70 216 142 143 144 331 333 160 266	O O O O O O O O O O

226

BINPU ROUTINE TO WRITE COL BIN CARDS ON TAPE. FIBII

																	156,							
																	153,							
! •																	150,							
DATA																	146,	335						
																	72,	7,						
BLY																	7	20						
ASSEMBLY								135									71,	203,						
POST PROCESSOR	173							132,									640	177,						
ST PRO	404			20		116		73,									63.	166,						
P 0	30,	140	4.5	20,		100,	25	41,	137		123	126	141	131	124	130	26,	162,	174		32	33	125	
)))	BLSEQ	BPTES	COSEQ	COUNT	10123	IDLCD	IMAGE	SEQNO	SWTCH	WRITE	(105)	(RCH)	(TES)	(WER)	(WRS)	(MTC)	CIMAGE	COMMON	COSEQX	MSK2CH	MSKPDT	MSKTSX	PUNCMD	WR I TE1
; ; ;	302	131	172	6.1	245	326	334	267	146	121	0	7	ß	4	-	m	140	_					m	

335

213,

146, 150, 153, 156, 164, 165, 170, 335

777740 77777 211 (211 (242 1) (242 1) (307 1) (307 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1) (313 1)

NO ERROR IN ABOVE ASSEMBLY.

10/11/62 MPUNCH01 MPUNCH02 MPUNCH03	R D	MPUNCH12 NTAPE)MPUNCH13 MPUNCH14 MPUNCH15	MPUNCH16 MPUNCH17 MPUNCH18 MPUNCH19 MPUNCH20	MPUNCH21 MPUNCH22 MPUNCH23 MPUNCH24	MPUNCH26 MPUNCH27 MPUNCH28 MPUNCH29 MPUNCH29	MPUNCH31 MPUNCH32 MPUNCH33 MPUNCH34 MPUNCH35 MPUNCH35 MPUNCH36 MPUNCH37
MPUN _{CH} C MATRIX COLUMN BINARY PUNCH SUBROUTINE C CALLING SEQUENCE	C A = MATRIX 3O BE PUNCHED IORG = ORIGIN OF FIRST CARD C M = NUMBER OF ROWS BCDZ = BCD IDENTIFICATION W C N = NUMBER OF COLUMNS MAXM = DIMENSIONED NUMBER O C IOUT = 0, PUNCH BY COLUMNS E 1, PUNCH BY ROWS NTAPE= OUTPUT PUNCH TAPE C ITRA = 0, TRA CARD AFTER WHOLE MATRIX C = 1, TRA CARD AFTER EACH ROW OR COLUMN	SUBROUTINE MPUNCH (A, M, N, IOUT, ITRA, IORG, BCDZ, MAXM, NT/ * LIST8 DIMENSION A(1), T(22)	IS = 0 MN = MAXM*N IF (10UT) 8,2 ;8		Ji- IONG DD 5 I=1,N CALL BINPU (A(J), M, Jl, BCDZ, IS, NTAPE) J = J+MAXM IS= IS+1+M/22 IF (ITRA) 3,4,3	C PUNCH TRA CARD AFTER EACH COLUMN 3 CALL BINPU (A, 0, 0, BCDZ, IS, NTAPE) 1S=IS+1 GOTO 5 4 J1=J1+M

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MPUNCH39
           MPUNCH40
                                                              MPUNCH45
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                                                                                                                                               MP UNCH53
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(T,22,41,8CDZ,IS,NTAPE) (T,J,J2,BCDZ,IS,NTAPE) 10,8,9 13,14613 K=I,MN,MAXM C PUNCH MATRIX BY ROWS.

14 I=1,M

00 00

J1=10RG

0**=**0 8

IF (J-22)

9 CALL BINPU

J1 = J1 + 22

IS=IS+1

0=0

T(J) = A(K)

1=1+1

10

(A, O, O, BCDZ, IS, NTAPE)

CALL BINPU GOTO 16

C PUNCH TRA GARD AFTER WHOLE MATRIX

(ITRA) 16,6\$16

ΙŁ

5 CONTINUE

MPUNCH

12,12,12 CALL BINPU CONTINUE 11=11+1 ΙŁ 10 71

(T,0,0,06BCDZ, IS,NTAPE) PUNCH TRA CARD AFTER EVERY ROW 13 CALL BINPU IS=IS+1 J1=10RG ں

12 IF (ITRA)

IS=IS+1

0=0

(ITRA) 16,15,16 AFTER ENTIRE MATRIX ٦I S, ں

MPUNCH78

-95-

10/11/62

MPUNCH79 MPUNCH80 MPUNCH81 MPUNCH82

END(1,1,0,0,0,0,0,1,0,1,0,0,0,0,0,0)

15 CALL BINPU (A,0,0,0BCDZ, IS, NTAPE)

16 RETURN

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STORAGE NOT USED BY PROGRAM

DEC 0CT 32561 77461

DEC 0CT 279 00427

TIONS FOR VARIABLES APPEARING IN DIMENSION AND EQUIVALENCE STATEMENTS	DEC OCT	DIMENSION, OR EQUIVALENCE STATEMENTS	DEC 0CT J 253 00375	I SOURCE PROGRAM	DEC OCT 9) 247 00367 E)H 215 00327	VECTOR	DEC OCT
DIMENSION	OCT		OCT 00376	PEARING IN	DEC 0CT 241 00361 120 00170	RANSFER VE	100
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ENTRY POINTS TO SUBROUTINES NOT OUTPUT FROM LIBRARY

EXTERNAL FORMULA NUMBERS WITH CORRESPONDING INTERNAL FORMULA NUMBERS AND OCTAL LOCATIONS IFN LOC 20 00166 37 00263 48 00330 10 IFN LDC 19 00163 32 00241 44 00312 EFN IFN LOC 15 00147 25 00205 43 00310 52 00352 EFN LOC 00105 00175 00271 00343 1FN 7 22 (39 0 50 00 EFN

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00 +223145476460 0	BCD 1BINPU	+062100000347 01	A 1A+17
00 0000000000000 10	PZE	+062100000316 01	A 1A+14
02 +000000000000 0	PZE	+062100000275	STA 1A+132
03 +00000000000000000000000000000000000	PZE	+062100000246 01	A 1A+10
04 -044764452330 0	BCD 1MPUNCH	+062100000201 01	A 14+7
05 ~063400100001 01	SXD \$,1	+062100000153 01	A 14+5
06 -063400200002 0	SXD \$+1,2	+062100000124 01	A 1A+2
07 -063400400003 01	XD \$+2,	+050000000010	A 8
10 +050000400001 00	LA 1,	+062100000223 01	A 1A+
11 +062100000344 01	TA 1,	+062100000130 0	A 1A+3
12 +040000000367 01	90	+062100000075 01	A 1A+
13 +062100000232 01	TA 1	+05000000011 0	A 9,
14 +040000000370 01	ADD 9)+1	+062100000351 01	A 1A+17
15 +062100000176 01	TA 1A+6	+062100000320 01	A 1A+1
16 +062100000150 01	TA 1A+	+062100000277 01	A 1A+13
17 +062100000121 01	TA 1A+2	+062100000250 01	A A
20 +040000000367 01	6 00	+062100000203 0	A 1A+7
21 +062100000115 01	TA 1A	+062100000155 01	A 1A
22 +050000400002 00	LA 2,	+062100000126 01	A 1A+2
23 +062100000214 01	TA 1A+8	+050000000002 0	4
24 +062100000164 0	TA 1A	+060100000366 01	STO 6145
25 +062100000133 01	TA 1A+3	+050000000356 0	⋖
26 +062100000122 01	TA 1A+2	+060100000377 01	<u>~</u> 0
27 +05000400003 00	LA 3,	÷05600000000000000	Σ Ø
30 +062100000113 01	TA 1A+	+02000000000000000	>
31 +062100000076 0	TA 1	+076700000021 0	S 1.
32 +05000400004 00	LA 4,4	+060100000374 0	NW O
33 +062100000101 0	TA 1	+0500000000000000000000000000000	A
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35 +062100000337 01	TA 1A+1	+012000000205 01	1 2
36 +062100000310 01	TA 1A+14	+002000000205 01	A 2
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10/11/62 PAGE 7	26 010 40A TSX B 26 010 10 40A TSX T TSX T TSX T TSX J TSX J TSX J TSX J TSX J TSX B 100 00 00 153 B	77 010	010 42A CLA 010 010 STD 010 43A CLA 010 43A1 TZE 144A BSS	010 TSX T 010 TSX 2) 010 TSX 2) 00 TSX 8CDZ 010 TSX 1S 010 TSX NTAPE 010 46A CLA IS 010 STO IS	000 47A 010 010 010 E)H 010 48A 010 010
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(WRS)	POSITIONING ERRORS	0	0	0	0	0	0	0
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(EFT)	TOTAL REDUNDANCIES WRITING READING	0	0	0	0	0	0	0
(FIL)	CORDS READING	0	0	0	0	0	0	0
RARY, (STHM)	NOISE RECORDS WRITING READI	0	0	0	0	0	0	0
TED FROM LIB (RWT) (TES)	TOTAL READS	522	554	102	405	635	٣	145
JTINES REQUES (RTN) (RTN)	TOTAL WRITES	0	536	91	372	0	793	141
ENTRY POINTS TO SUBROUTINES REQUESTED FROM LIBRARY, (FPT) (TSHM) (RIN) (RWT) (RCH) (WTC) (WER) (TES)	MACHINE TAPE	. A	8 2	R 80	7 V	A 2	A 3	8 4

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Aerospace Corporation, El Segundo, California. AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY: ANALYTICAL DEVELOPMENT AND COMPUTATIONAL PROCEDURE, prepared by W. P. Rodden, E. F. Farkas, and G. Y. Takata. 31 October 1962. [107]P. incl. illus. (Report TDR-1693230-11)TN-6;SSD-TDR-62-149) (Contract AF 04(695)-169)		Aerospace Corporation, El Segundo, California. AERODYNAMICS INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY: ANALYTICAL DEVELOPMENT AND COMPUTATIONAL PROCEDURE, prepared by W. P. Rodden, E. F. Farkas, and G. Y. Takata. 31 October 1962. [107]p. incl. illus. (Report TDR-169(3230-11)TN-6; SSD-TDR-62-149) (Contract AF 04(695)-169)
A method is reviewed for computing the aerodynamic influence coefficients (AICs) for slender bodies. The method is based on the unsteady slender-body theory by Miles and its extension to obtain the AICs by Rodden and Revell. The simplicity of the slender-body theory permits the definition of a number of sets of AICs for use in transient analysis. The influence coefficients relating the transient aerodynamic forces to the body deflections and their first two derivatives are defined by the		A method is reviewed for computing the aerodynamic influence coefficients (AICs) for slender bodies. The method is based on the unsteady slender-body theory by Miles and its extension to obtain the AICs by Rodden and Revell. The simplicity of the slender-body theory permits the definition of a number of sets of AICs for use in transient analysis. The influence coefficients relating the transient aerodynamic forces to the body deflections and their first two derivatives are defined by the
following relation: (over)	UNCLASSIFIED	following relation: (over)
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Aerospace Corporation, El'Segundo, California. AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY: ANALYTICAL DEVELOPMENT AND COMPUTATIONAL PROCEDURE, prepared by W. P. Rodden, E. F. F. F. P. Rodden, F. F. F. F. P. Rodden,		Aerospace Corporation, El Segundo, California. AERODYNAMIC INFLUENCE COEFFICIENTS FROM SLENDER-BODY THEORY: ANALYTICAL DEVELOPMENT AND COMPUTATIONAL PROCEDURE, Prepared by W. P. Rodden, E. F. Farkas, and G. Y. Takata.
107 p. incl. illus. 107 p. incl. illus. Report TDR-169(3230-11)TN-6; SSD-TDR-62-149) Contract AF 04(695)-169)		[107]p. incl. illus. (Report TDR-169(3230-11)TN-6; SSD-TDR-62-149) (Contract AF 04(695)-169) Unclassified report
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UNCLASSIFIED Corporation, El Segundo, California.
CINFLUENCE COEFFICIENTS
ER-BODY THEORY: ANALYTICAL
AND COMPUTATIONAL
prepared by W. P. Rodden,
and G. Y. Takata. 31 October 1962. wed for computing the aerodynamic transient aerodynamic forces to the body deflections and their first two derivatives are defined by the 9(3230-11)TN-6; SSD-TDR-62-149) (695)-169) Unclassified report theory by Miles and its extension to obtain the AIGs by Rodden and Revell. The simplicity of the slender-body theory permits the definition of a number of sets of AIGs for use in transient analysis. The influence coefficients relating the The method is based on the unsteady slender-body influence coefficients (AICs) for slender bodies. (over) following relation:

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transient aerodynamic forces to the body deflections and their first two derivatives are defined by the

(over)

following relation:

slender-body theory permits the definition of a number of sets of AICs for use in transient analysis. The influence coefficients relating the

The method is based on the unsteady slender-body theory by Miles and its extension to obtain the AICs by Rodden and Revell. The simplicity of the

	$\begin{split} F(t) &= (q S/\overline{c}) \left\{ \left[C_{h_S} \right] h \right\} + \left[C_{h_d} \right] h \overline{c} / V \right\} \\ &+ \left[C_{h_1} \right] \left\{ \overline{h} \overline{c}^2 / V^2 \right\} \right]. \end{split}$ The matrices C_{h_S} , C_{h_d} , and C_{h_1} are seen to be steady, damping, and inertial AICs, respectively. The oscillatory AICs are defined by $\{F\} = \rho \omega^2 \overline{b}_z \{C_h\} \} h \}$ and are related to the above definitions through $2k_z^2 (\overline{c}s/S) \{C_h\} = \{C_{h_S}\} + i k_z (\overline{c}/b_z) \{C_{h_d}\} \\ - k_z^2 (\overline{c}/b_z)^2 \{C_{h_1}\} \\ - k_z^2 (\overline{c}/b_z)^2 \{C_{h_1}\} \end{split}$ The Aerospace IBM 7090 Computer Program No. HMI5 provides the AICs in printed and optional purc.edcard output formats. The program of the above the AICs in printed and optional purc.edcard output formats. The program oscillatory case, 50 values of reduced velocity			$\{F(t)\} = (qS/\overline{c}) \left\{ [C_{hs}] \{h\} + [C_{hd}] \{h\overline{c}/v\} + [C_{hd}] \{h\overline{c}^2/v^2\} \right\}$	The matrices $\{C_{hs}\}$, $\{C_{hd}\}$, and $\{C_{hi}\}$ are seen to be steady, damping, and inertial AICs, respectively. The oscillatory AICs are defined by $\{F\} = \rho \omega^2 b_F^2 \{C_h\} \{h\}$	and are related to the above definitions through $2k_r^2(\sigma_s/S)$ [C_h] = [C_{hs}] + ik _r (σ /b _r) [C_{hd}]	$- \kappa_{\rm r}^2 (\bar{c}/b_{\rm r})^2 [C_{\rm hi}]$.	The Aerospace IBM 7090 Computer Program No. HM15 provides the AICs in printed and optional punched-card output formats. The pro- gram capacity is 50 control points and, in the oscillatory case, 50 values of reduced velocity
UNCLASSIFIED		UNCLASSIFIED	UNCLASSIFIED					UNCLASSIFIED
	$\{F(t)\} = (qS/\sigma) \{ C_{hs} \} \{h\} + \{ C_{hd} \} \{h\tau/v\} \} + \{C_{hi} \} \{h\sigma^2/v\} \}$ The matrices $\{C_{hs}\}$, $\{C_{hd}\}$, and $\{C_{hi}\}$ are seen to be steady, damping, and inertial AIGs, respectively. The oscillatory AIGs are defined by $\{F\} = \rho \omega^2 b_F^2 \{ C_h \} \{h\} \}$ and are related to the above definitions through $\{F\} = \rho \omega^2 b_F^2 \{ C_h \} \{h\} \}$. The Aerospace IBM 7090 Computer Program No. HMI5 provides the AIGs in printed and optional punched-card output formats. The program capacity is 50 control points and. in the oscillatory case, 50 values of reduced velocity.			$\{F(t)\} = (qS/\overline{c}) \left([C_{hs}] \{h\} + [C_{hd}] \} h\overline{c}/v \right\} + [C_{hi}] \{\overline{h}\overline{c}^2/v^2\} \right) .$	The matrices [C_{hs}], [C_{hd}], and [C_{hi}] are seen to be steady, damping, and inertial AICs, respectively. The oscillatory AICs are defined by $\{F\} = \rho \omega^2 b_r^2 \{C_h\}\{h\}$	and are related to the above definitions through $2k_r^2(\overline{c}s/S) \left[C_h\right] = \left[C_{hs}\right] + ik_r(\overline{c}/b_r) \left[C_{hd}\right]$	$- \kappa_{\rm r}^2 ({\it c}/{\it b}_{\rm r})^2 [{\it c}_{ m hi}]$.	The Aerospace IBM 7090 Computer Program No. HM15 provides the AICs in printed and optional punched-card output formats. The pro- gram capacity is 50 control points and, in the oscillatory case, 50 values of reduced velocity.

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	$\{F(t)\} = (qS/\overline{c}) \Big\{ \Big[C_{hS} \Big] \Big\{ h \Big\} + \Big[C_{hd} \Big] \Big\{ \overline{h}\overline{c}/v^2 \Big\} \Big\} \\ + \Big[C_{hi} \Big] \Big\{ \overline{h}\overline{c}/v^2 \Big\} \Big\}.$ The matrices $\Big[C_{hS} \Big]$, $\Big[C_{hd} \Big]$, and $\Big[C_{hi} \Big]$ are seen to be steady, damping, and inertial AICs, respectively. The oscillatory AICs are defined by $ \Big\{ F \Big\} = \rho \omega^2 b_b^2 s \Big[C_h \Big] \Big\{ h \Big\} $ and are related to the above definitions through $ Z_k^2 (\overline{c}s/S) \Big[C_h \Big] = \Big[C_{hS} \Big] + ik_r (\overline{c}/b_r) \Big[C_{hd} \Big] \\ - k_r^2 (\overline{c}/b_r)^2 \Big[C_{hi} \Big] $ The Aerospace IBM 7090 Computer Program No. HMI5 provides the AICs in printed and optional punched: card output formats. The program oscillatory case, 50 values of reduced velocity.	
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	$\label{eq:formula} \begin{split} \big\{F(t)\big\} &= (q_S/\sigma) \Big\{ \big[C_{h_S} \big] \big\{ h\big\} + \big[C_{h_d} \big] \big\{ h\sigma/v^{\big} \big\} \\ &+ \big[C_{h_1} \big] \big\{ h\sigma^2/v^2 \big\} \Big\} \; . \end{split}$ The matrices $[C_{h_S} \big]$, $[C_{h_d} \big]$, and $[C_{h_1} \big]$ are seen to be steady, damping, and inertial AIGs. respectively. The oscillatory AIGs are defined by $ \big\{ F \big\} = \rho \omega^2 b_T^2 g C_{h_1} \big\{ h \big\} \\ \text{and are related to the above definitions through} \\ 2k_T^2 (\overline{c}s/S) \big[C_{h_1} \big] + ik_T (\overline{c}/b_T) \big[C_{h_d} \big] \\ - k_T^2 (\overline{c}/b_T)^2 \big[C_{h_1} \big] \; . \end{split}$ The Aerospace IBM 7090 Computer Program No. HMI5 provides the AIGs in printed and optional punched-card output formats. The program capacity is 50 control points and, in the oscillatory case, 50 values of reduced velocity.	

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	$ \{F(t)\} = (9S/\overline{c}) \Big([C_{hs}] \{h\} + [C_{hd}] \{h\overline{c}/v\} \\ + [C_{hi}] \{h\overline{c}^2/v^2\} \Big) \ . $ The matrices $[C_{hs}]$, $[C_{hd}]$, and $[C_{hi}]$ are seen to be steady, damping, and inertial AICs, respectively. The oscillatory AICs are defined by $ \{F\} = \rho \omega^2 b_{rs}^2 [C_h] \{h\} $	and are related to the above definitions through $2k_{\rm r}^2({\tt Ts/S}) \left[C_{\rm h} \right] = \left[C_{\rm hs} \right] + ik_{\rm r} \left({\tt T/b_{\rm r}} \right) \left[C_{\rm hd} \right]$ $- k_{\rm r}^2 \left({\tt T/b_{\rm r}} \right)^2 \left[C_{\rm hi} \right]$ The Aerospace IBM 7090 Computer Program No. HMI5 provides the AIGs in printed and optional punched-card output formats. The program capacity is 50 control points and, in the oscillatory case, 50 values of reduced velocity
UNCLASSIFIED		UNCLASSIFIED
	$\begin{split} \left\{F(t)\right\} &= (qS/\overline{c}) \left\{ \left[C_{hg} \right] \left\{ h \right\} + \left[C_{hd} \right] \left\{ h\overline{c}^2/v^2 \right\} \right. \\ &+ \left[C_{hi} \right] \left\{ \overline{h}\overline{c}^2/v^2 \right\} \right. \end{split}$ The matrices $\left\{ C_{hg} \right\}$, $\left\{ C_{hd} \right\}$, and $\left\{ C_{hi} \right\}$ are seen to be steady, damping, and inertial AIGs, respectively. The oscillatory AIGs are defined by $\left\{ F \right\} = \rho \omega^2 b_{\mu}^2 \left\{ C_{hi} \right\} \left\{ h \right\} \end{split}$	and are related to the above definitions through $2k_{\mathbf{r}}^2(\overline{c}_{\mathbf{s}}/S)\left[C_{\mathbf{h}}\right] = \left[C_{\mathbf{h}_{\mathbf{S}}}\right] + ik_{\mathbf{r}}\left(\overline{c}/b_{\mathbf{r}}\right)\left[C_{\mathbf{h}_{\mathbf{d}}}\right] \\ - k_{\mathbf{r}}^2\left(\overline{c}/b_{\mathbf{r}}\right)^2\left[C_{\mathbf{h}_{\mathbf{l}}}\right] ,$ The Aerospace IBM 7090 Computer Program No. HM15 provides the AICs in printed and optional punched-card output formats. The program appacity is 50 control points and, in the oscillatory case, 50 values of reduced velocity.